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MEMORANDUM FOR PRR (~~Contractor~~/In-House Publication)

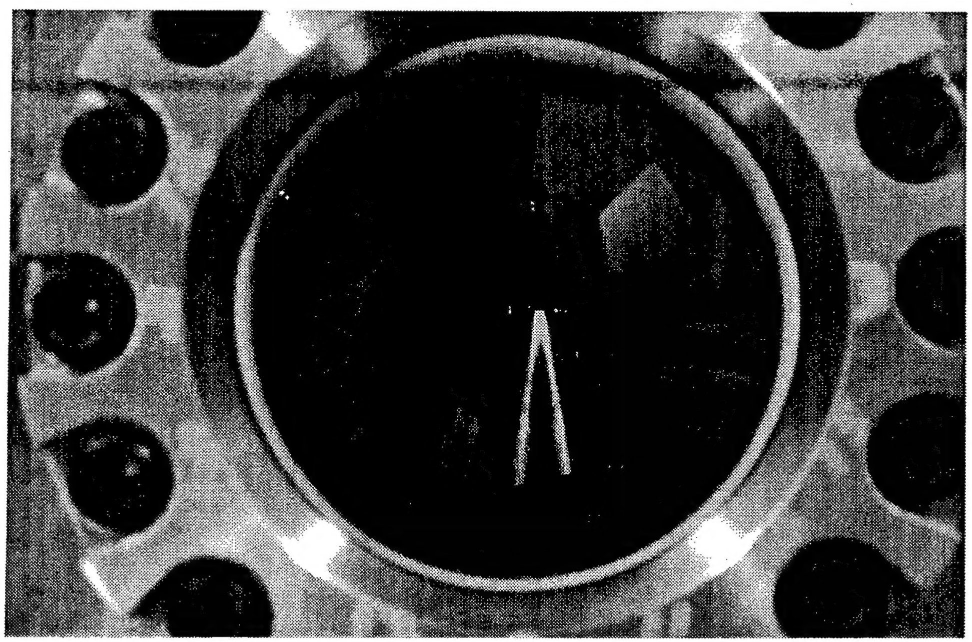
FROM: PROI (TI) (STINFO) 18 May 1999

SUBJECT: Authorization for Release of Technical Information, Control Number: AFRL-PR-ED-TP-FY99-0099
Doug Talley, "Basic Research in Liquid Rocket Combustion at the Air Force Research Laboratory"

International presentation **(Statement A)**

INTERNATIONAL WORKSHOP ON
RESEARCH STATUS AND PERSPECTIVES IN
LIQUID ROCKET COMBUSTION CHAMBER FLOW DYNAMICS

Basic Research in Liquid Rocket Combustion at the Air Force Research Laboratory



Doug Talley
Propulsion Sciences and Advanced Concepts Division

27-28 May 1999

Integrated High Payoff Rocket Propulsion Technology Program (IHPRPT) GOALS

Boost and Orbit Transfer Propulsion	2000	2005	2010
• Reduce Stage Failure Rate	25%	50%	75%
• Improve Mass Fraction (Solids)	15%	25%	35%
• Improve ISP (sec)	14	21	26
• Reduce Hardware Costs	15%	25%	35%
• Reduce Support Costs	15%	25%	35%
• Improve Thrust to Weight (Liquids)	30%	60%	100%
• Mean Time Between Removal (Mission Life-Reusable)	20	40	100
Spacecraft Propulsion			
• Improve I_{tot}/Mass $_{(wet)}$ (Electrostatic/Electromagnetic)	20%/200%	35%/500%	75%/1250%
• Improve Isp (Bipropellant/Solar Thermal)	5%/10%	10%/15%	20%/20%
• Improve Density-Isp (Monopropellant)	30%	50%	70%
• Improve Mass Fraction (Solar Thermal)	15%	25%	35%
Tactical Propulsion			
• Improve Delivered Energy	3%	7%	15%
• Improve Mass Fraction (Without TVC/Throttling)	2%	5%	10%
• Improve Mass Fraction (With TVC/Throttling)	10%	20%	30%

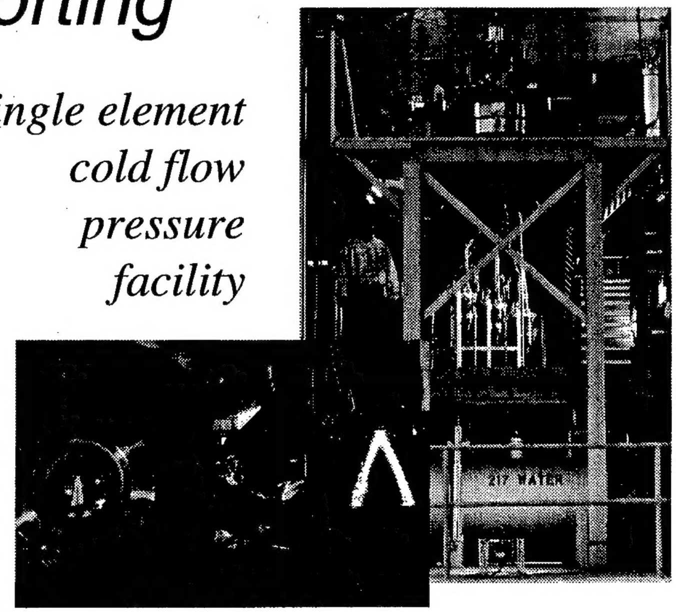
Required Injector Characteristics

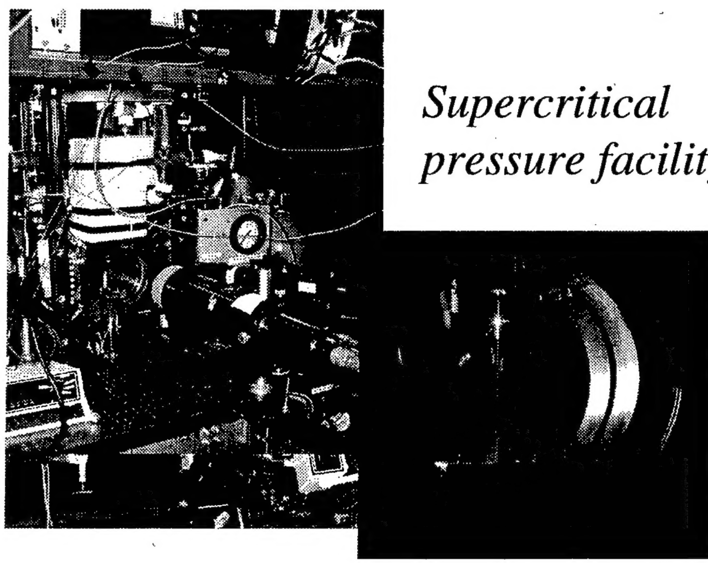
- Complete combustion in the shortest possible length
  - Main injectors: performance *vs* weight tradeoffs
  - Preburners/GG's: downstream component interactions, eg, turbine blades, etc
- Acoustically stable
  - Chamber modes
  - Feed system coupling
- Chamber/wall compatibility
  - Heat transfer/cooling
  - Oxygen blanching
- Minimize pressure drop
- Throttling
- Ignitable; minimum ignition transients
- Cost, weight
- The "ilities:"
  - Reliability
  - Maintainability
  - Manufacturability
  - Durability
  - Operability

Facilities currently supporting basic research

- All facilities designed to provide optical access at pressures to 2000 psi (136 atm).

Single element cold flow pressure facility



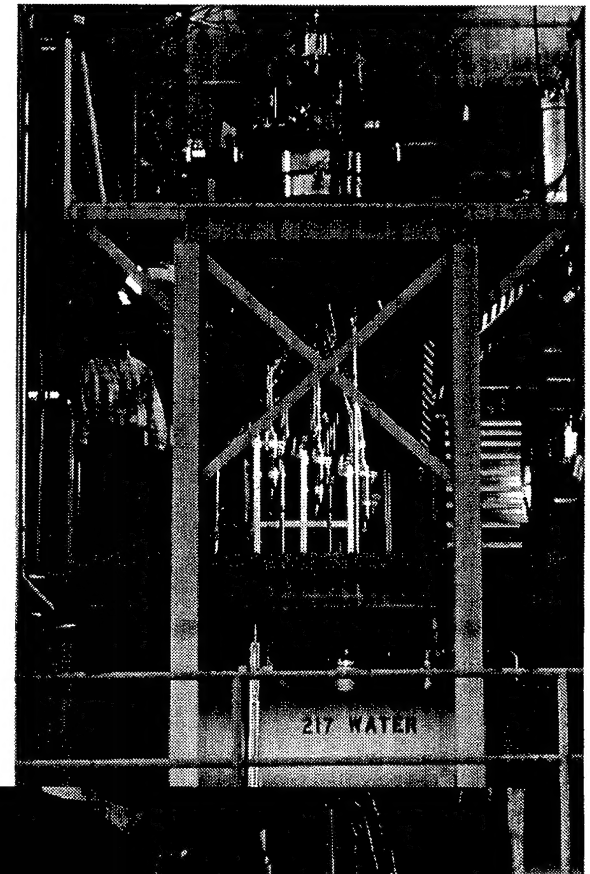


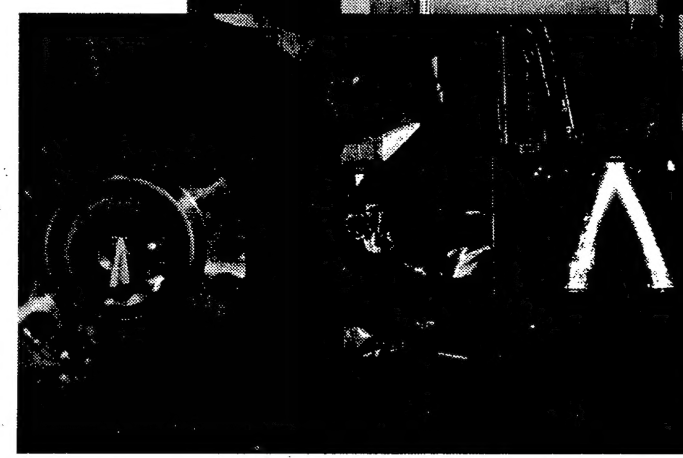
Supercritical pressure facility

Subscale hot fire facility

Single element cold flow pressure facility

Gas simulants	N_2(g), He(g)
Liquid simulant	H_2O(l), others
Window Purge gas	N_2(g), He(g)
N_2 mass flow rate	.20 lbm/s (.09 Kg/s)
He mass flow rate	.20 lbm/s (.09 Kg/s)
H_2O mass flow rate	4.0 lbm/s (1.8 Kg/s)
Max. test art. press.	2000 psi (136 atm)
Max. Fuel sim. press.	3000 psi. (204 atm)
Max. Ox sim. press.	3000 psi. (204 atm)





Windowed test chamber with 5.5" (14 cm) of axial injector travel and a linear translating injector stage with 5" (13 cm) total radial travel inside chamber.

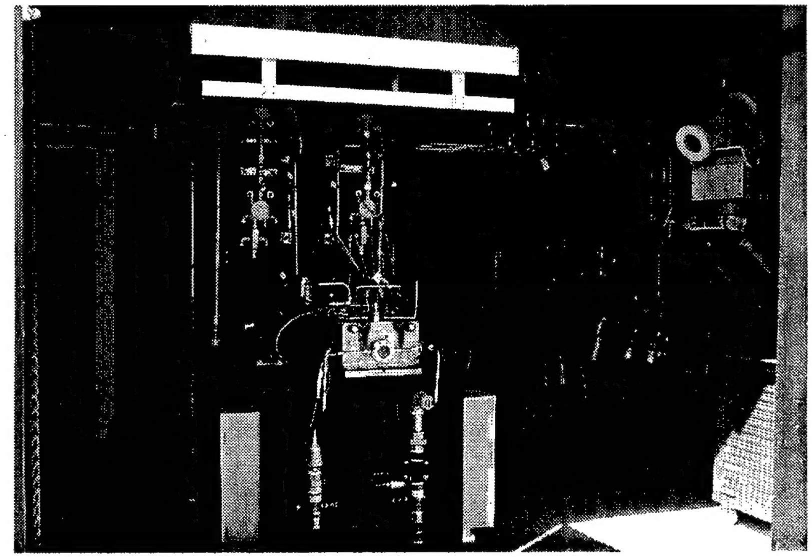
Ability to simulate manifold cross velocities to 30 ft/s (9.1 m/s).

27 tube traversable mechanical patternator

Phase Doppler, Malvern, other diagnostics

Subscale hot fire facility

Fuel	$H_2(g)$, $CH_4(g)$
Oxidizer	$O_2(g)$
Purge gas	$N_2(g)$, He(g)
H2 mass flow rate	.15 lbm/s (.07 Kg/s)
CH4 mass flow rate	.25 lbm/s (.11 Kg/s)
O2 mass flow rate	1.0 lbm/s (.45 Kg/s)
N2 mass flow rate	.5 lbm/s (.23 Kg/s)
Water flow rate	16 lbm/s (7 Kg/s)
Max. system press.	2640 psi. (179 atm)



128 ch, 200 kbs scanning A/D

16 ch, 2 MHz per ch A/D, independently controlled

Central laser/optics room

Installing in 1999

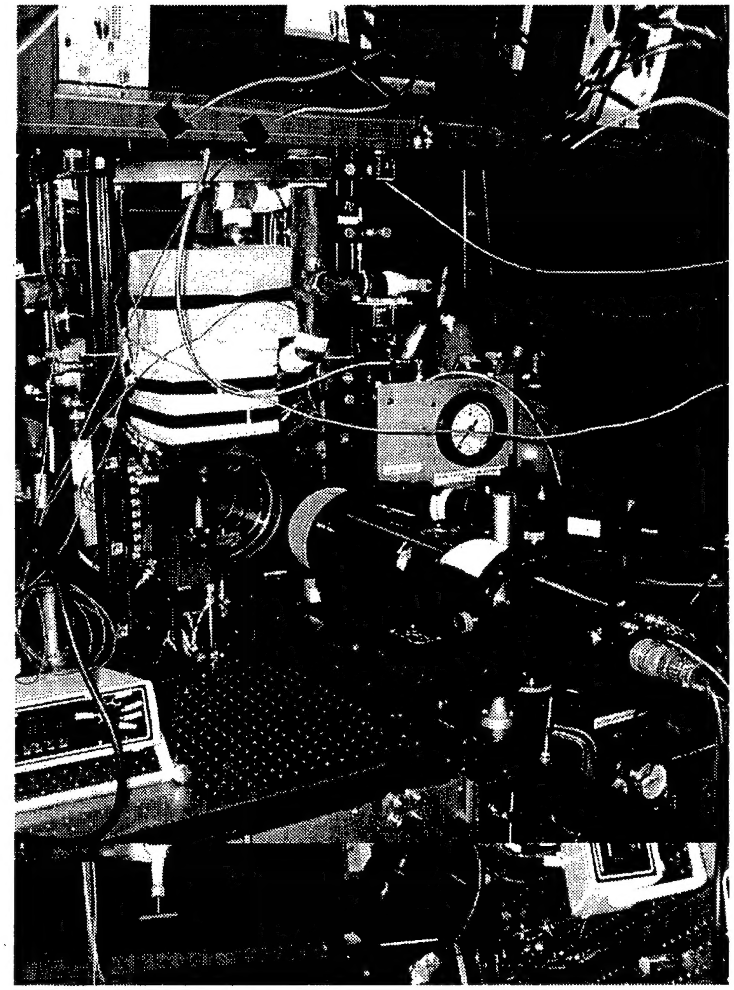
2000 psi liquid hydrocarbon capability

2000

LOX capability

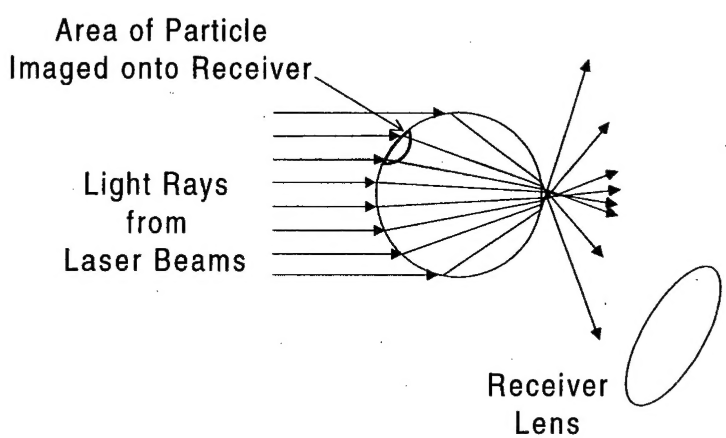
Supercritical pressure facility

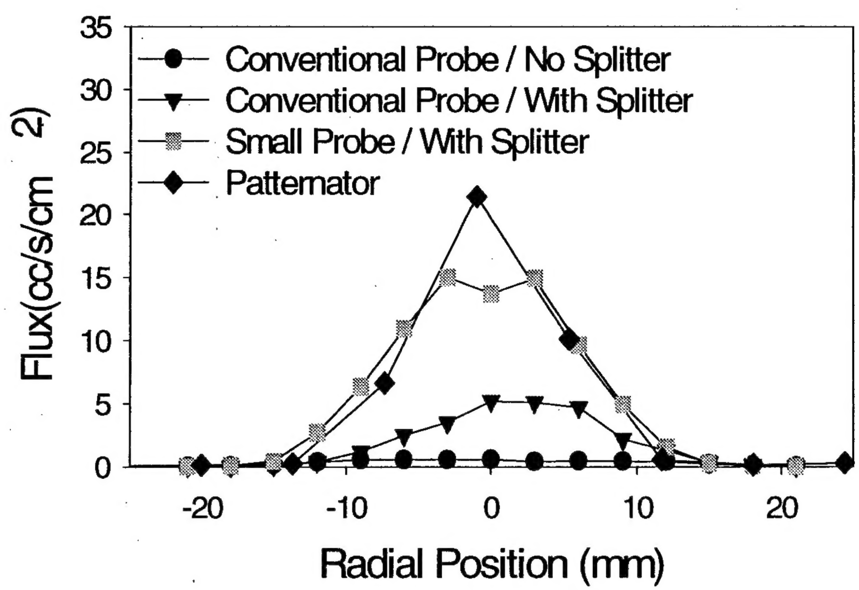
Chamber	Stainless
Optical access	2 facing sapphire 13 cm dia windows
	2 facing slot-shaped quartz (12 x 1.3 cm)
Max chamb. press.	2000 psi (136 atm)
Chamb. temp.	473 K
Injected fluid	O2, N2, HC, and mixtures
Ambient fluid	N2, He, and mixtures
Injected mass flow rate	400 mg/s
Cryogenic cooler	85 K
Mass flow meters	up to 10,000 SLPM



Dense Spray Diagnostics

- Goal - Extend existing diagnostic techniques into the dense spray regime where $N > 10^5\ cc^{-1}$.
- The combination of a small probe volume and a flow splitter resulted in a dramatic improvement in PDPA volume flux measurements in a dense spray.





Effect of Crossflow on Orifice Discharge Coefficients

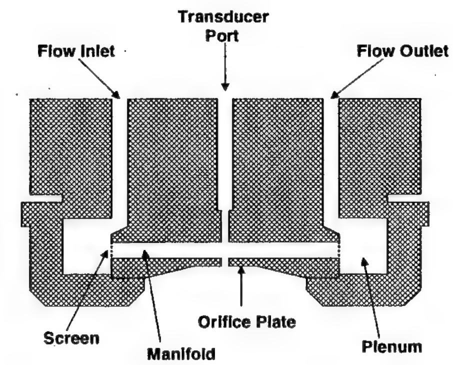
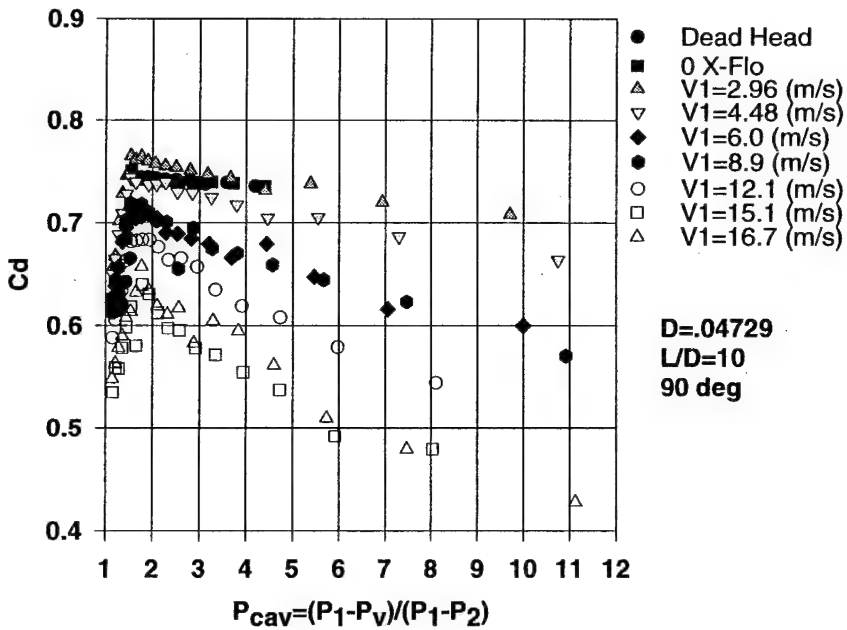
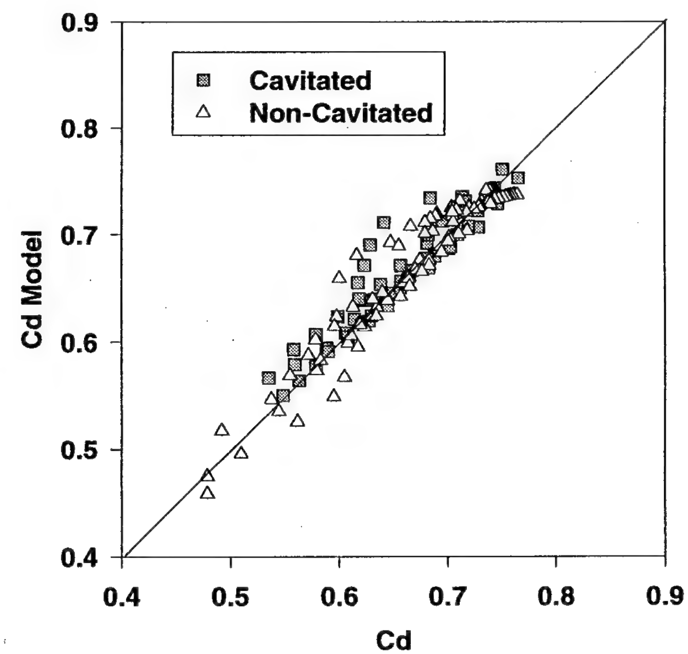


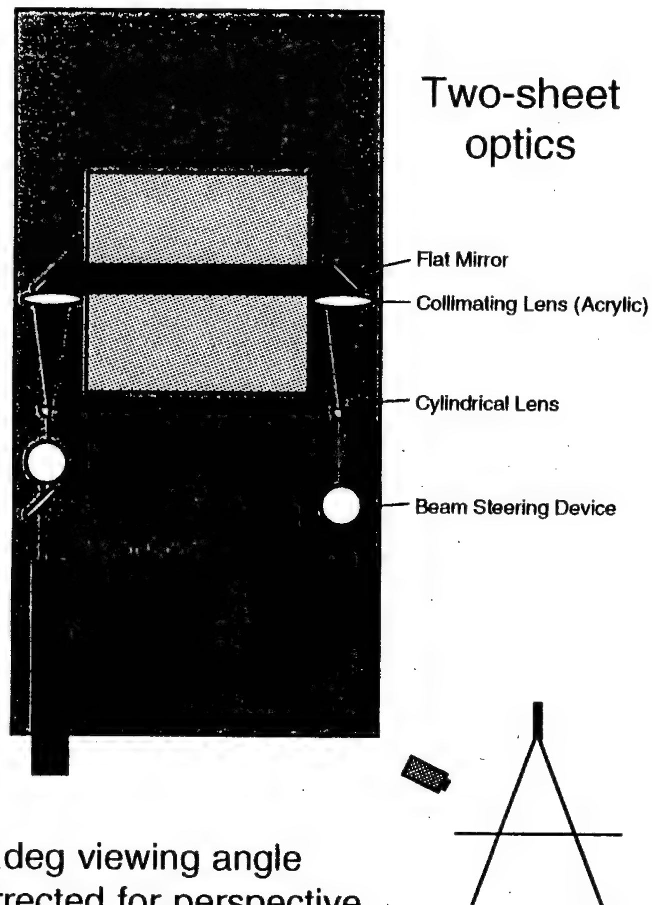
Figure 1





D=1.19 mm, L/D=10, P_1=.69 MPa

TWO SHEET CORRECTION FOR LASER EXTINCTION



20 deg viewing angle
corrected for perspective

Camera response*

$$G = KI\rho$$

Along a ray,

$$\frac{dG_l}{G_l} = \frac{dI_l}{I_l} + \frac{d\rho}{\rho}$$

$$\frac{dG_r}{G_r} = \frac{dI_r}{I_r} + \frac{d\rho}{\rho}$$

Beer's law scattering is the same in both directions at any *dx*. Thus

$$dI_l / I_l = -dI_r / I_r$$

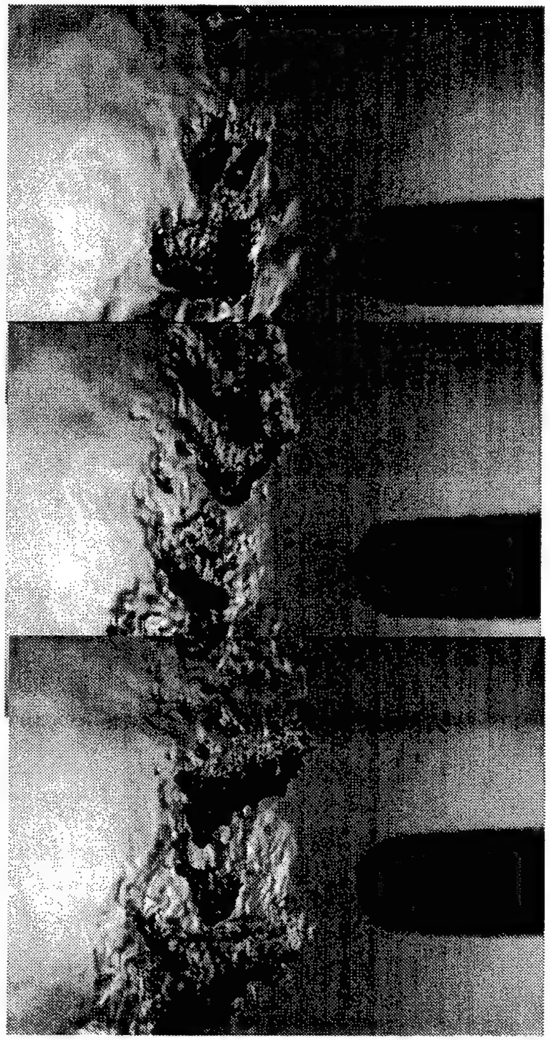
Adding,

$$\frac{dG_l}{G_l} + \frac{dG_r}{G_r} = 2\frac{d\rho}{\rho}$$

Independent of laser sheet intensity.

*G is gray level, I is laser sheet intensity, ρ is spray mass density, and K is a constant

High Pressure and Supercritical Combustion



Transcritical Oxygen Drops in Nitrogen

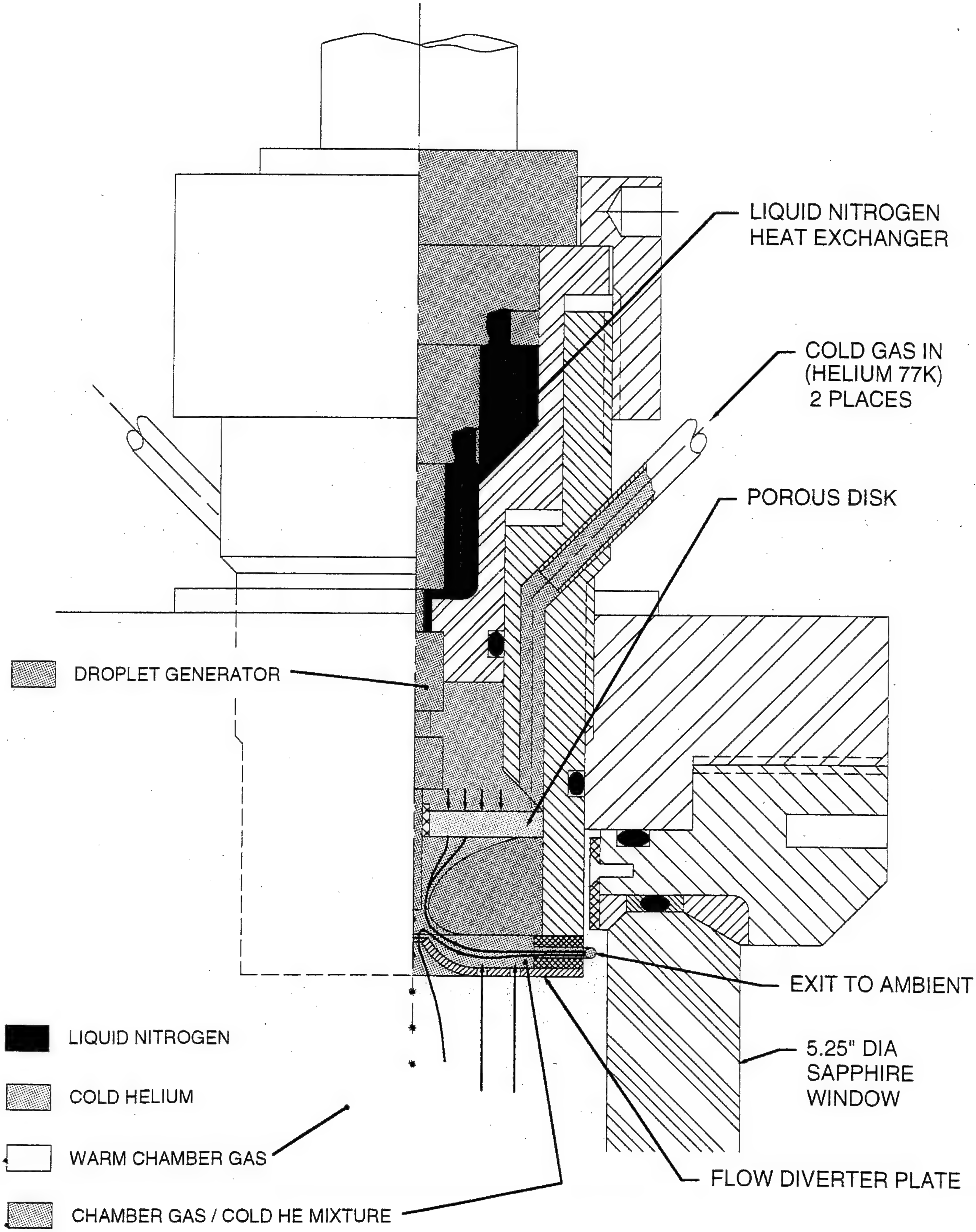
OBJECTIVE

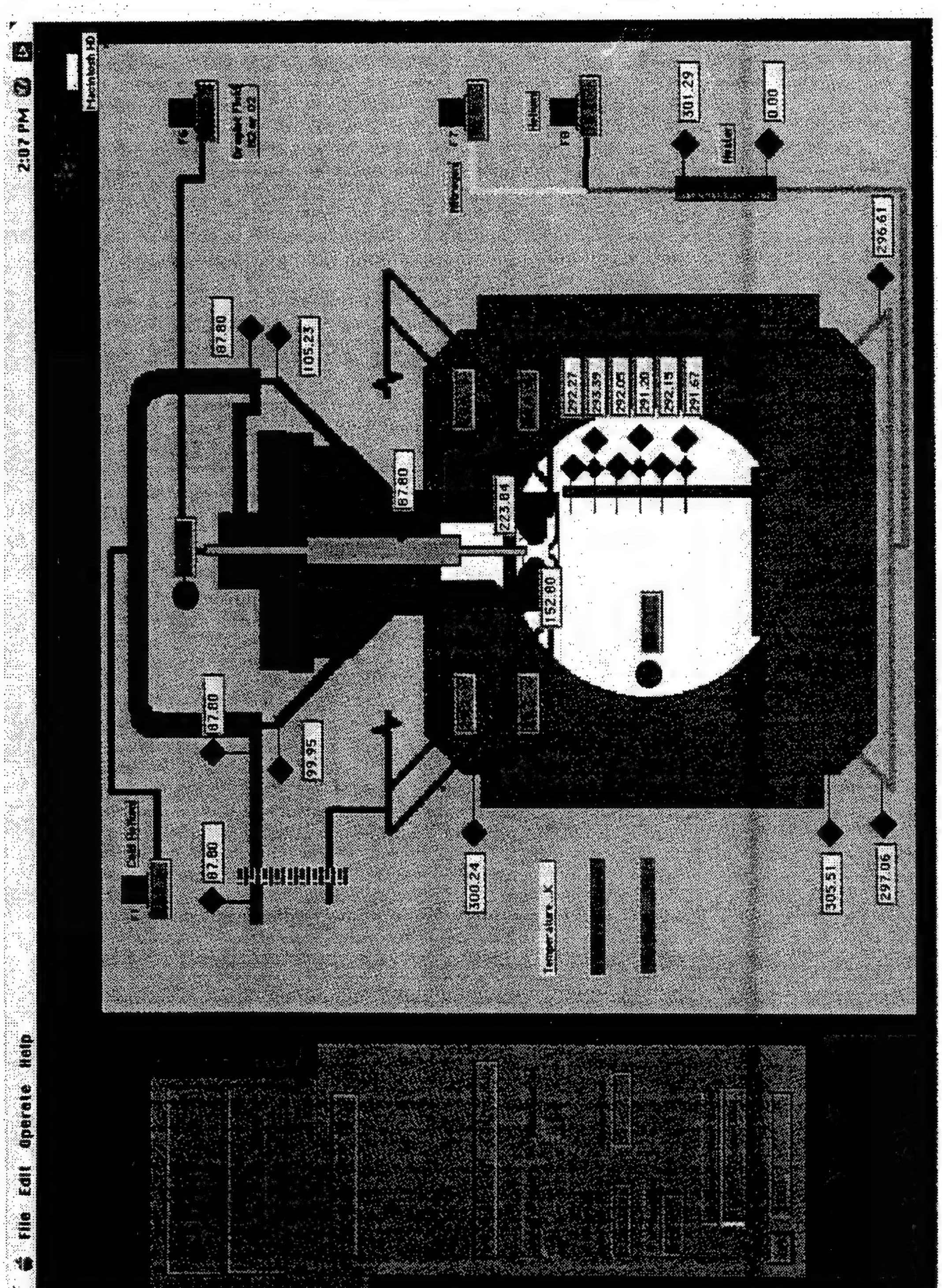
Determine the mechanisms which control the breakup, transport, mixing, and combustion of high pressure and supercritical droplets, jets, and sprays.

APPROACH

- Working fluids include cryogenics
  - LOX, LN2
- Droplet studies use *free* droplets to allow realistic deformation and breakup
- Shadowgraph, Schlieren, and planar Raman visualization of concentrations
  - PDPA, Malvern for subcrititical cases
- Counter dense sprays/distortions by reducing optical path lengths.

PRESSURE VESSEL / DROPLET GENERATOR INTERFACE





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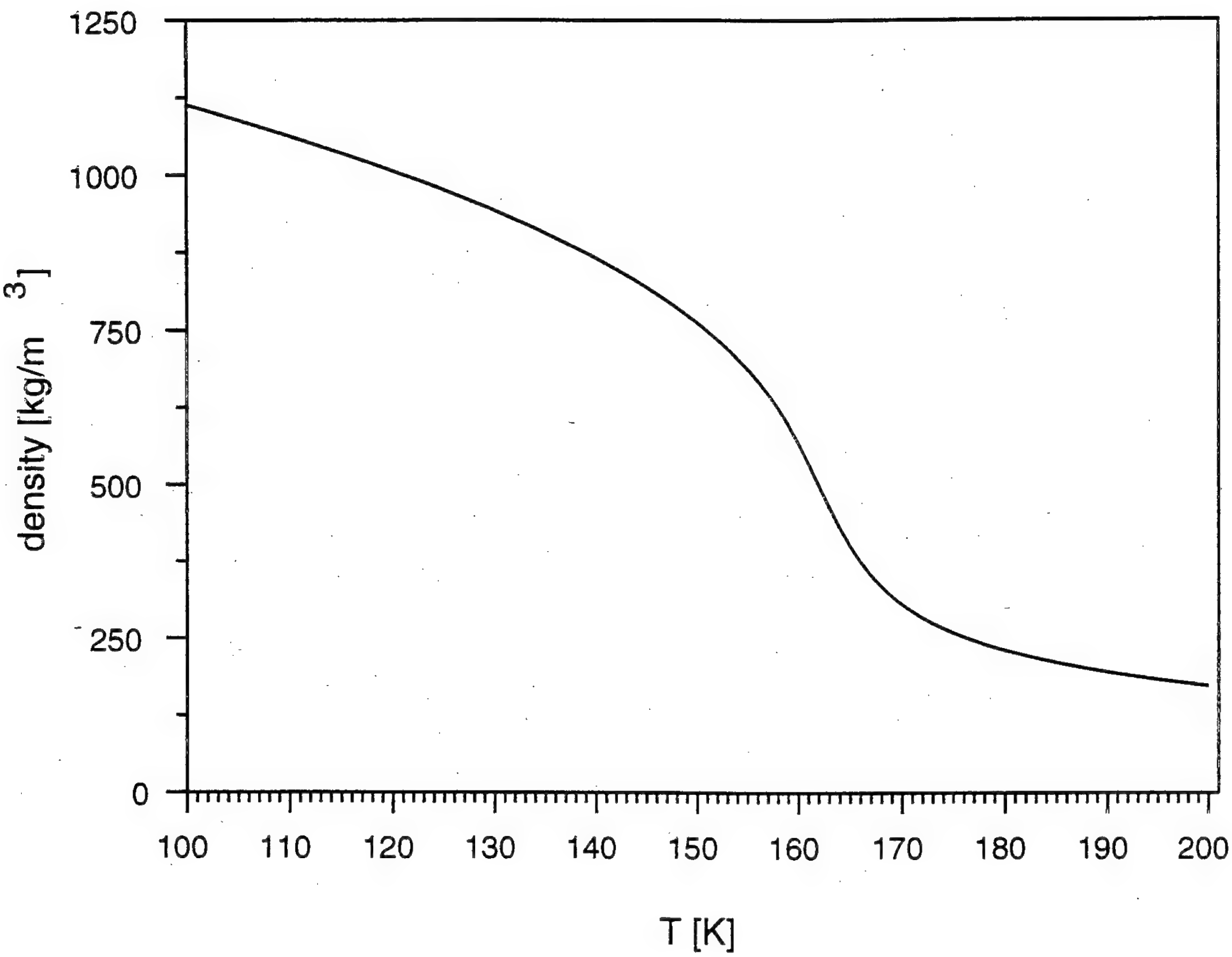
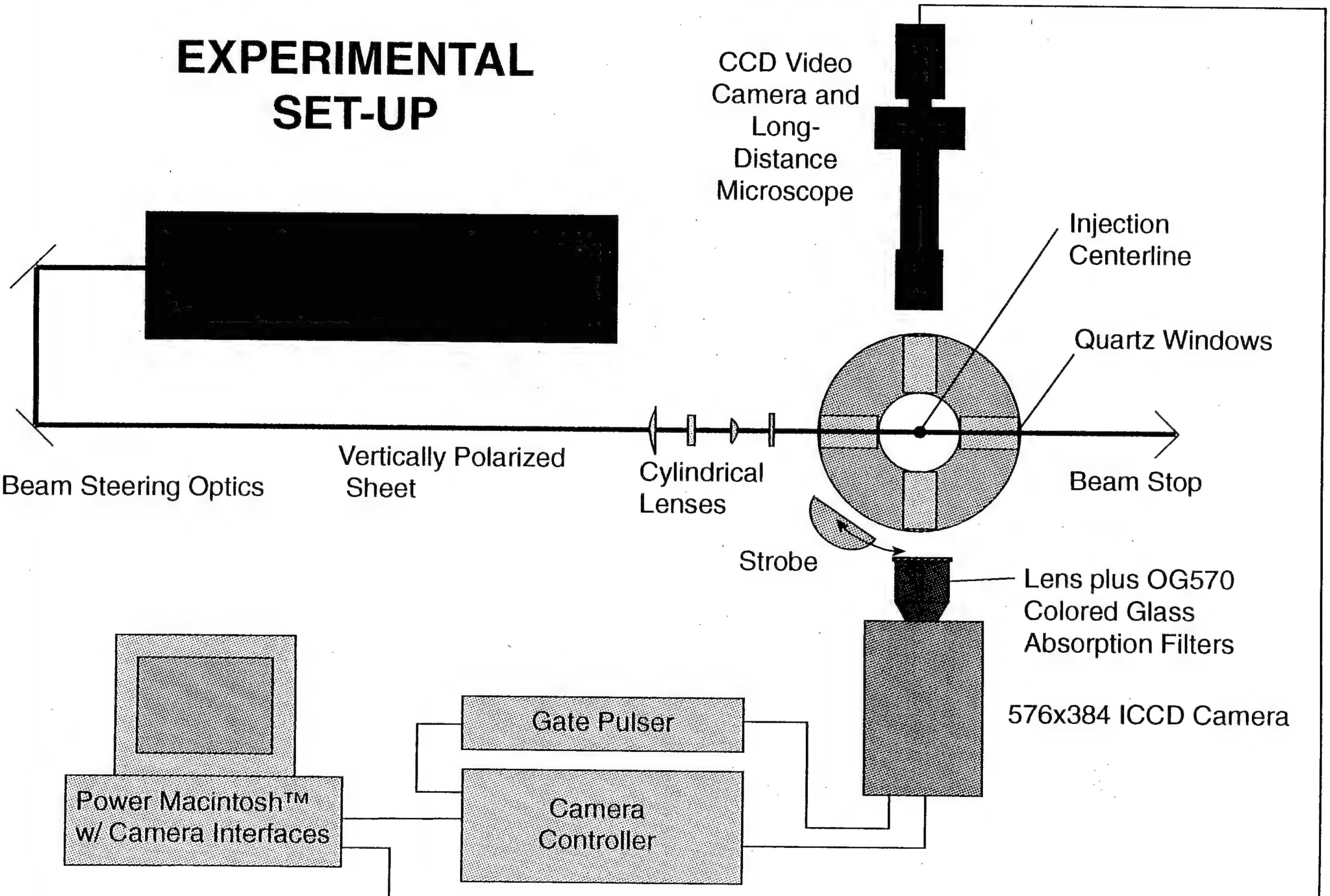
Oxygen Density at 69 atm



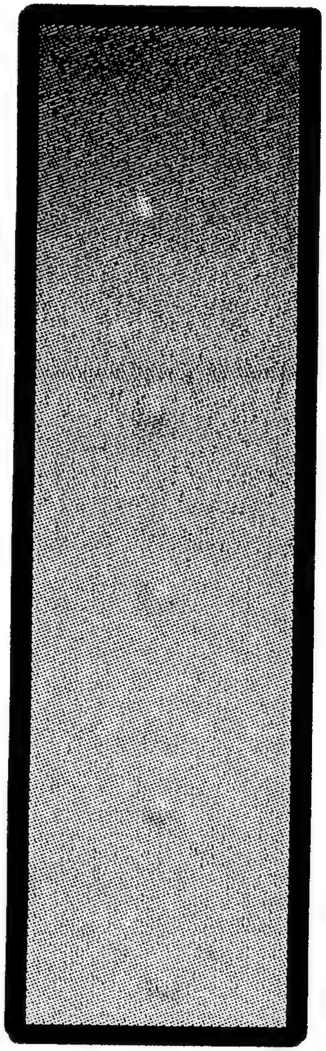
TRANSCRITICAL PROPELLANT INJECTION EXPERIMENTS

EXPERIMENTAL SET-UP

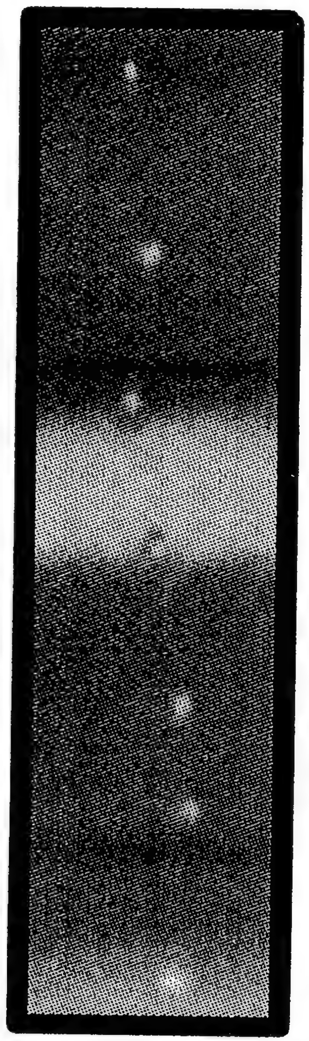


SUPERCRITICAL DROPLET BEHAVIOR
LOX DROPLETS INTO HELIUM

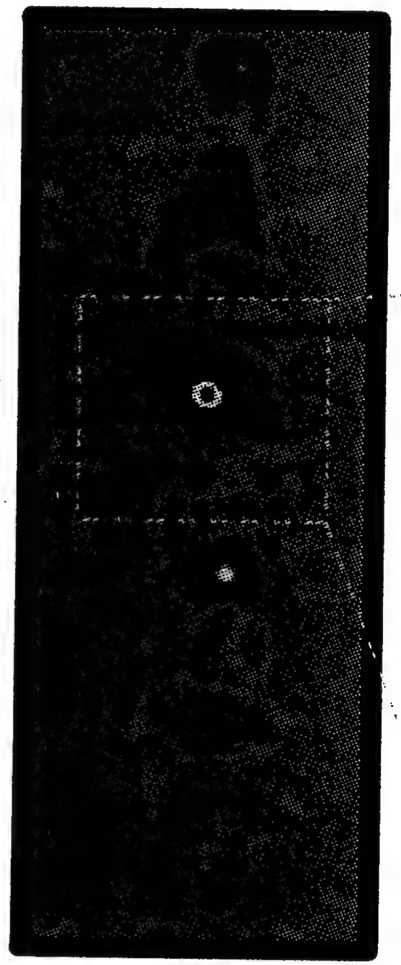
(a) 200 psig



(b) 1000 psig



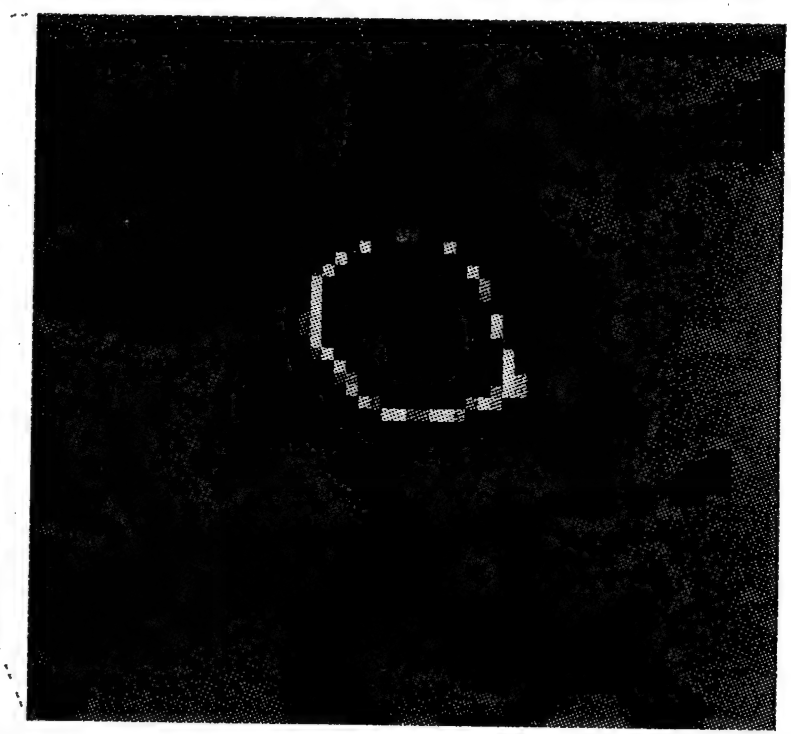
(c) 1000 psig



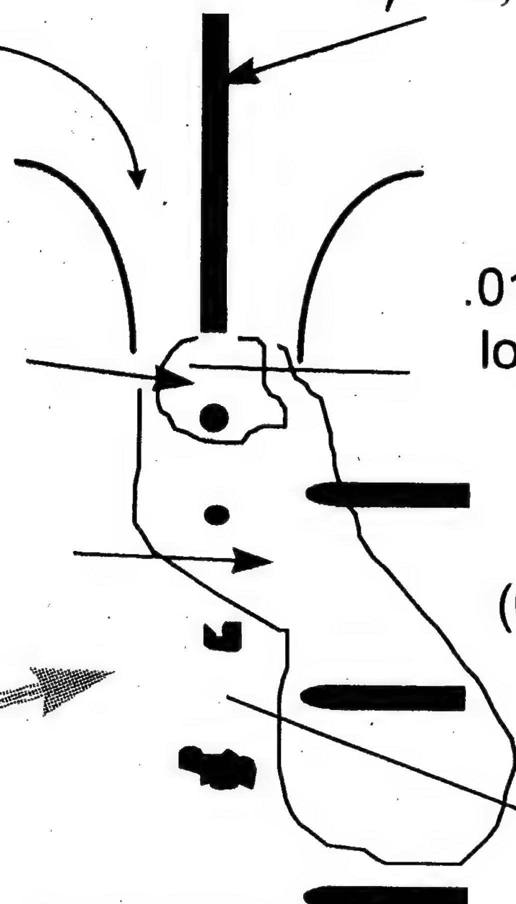
DROP SIZE ~ 200 μm

(a) and (b): Photographs indicate effect of pressure

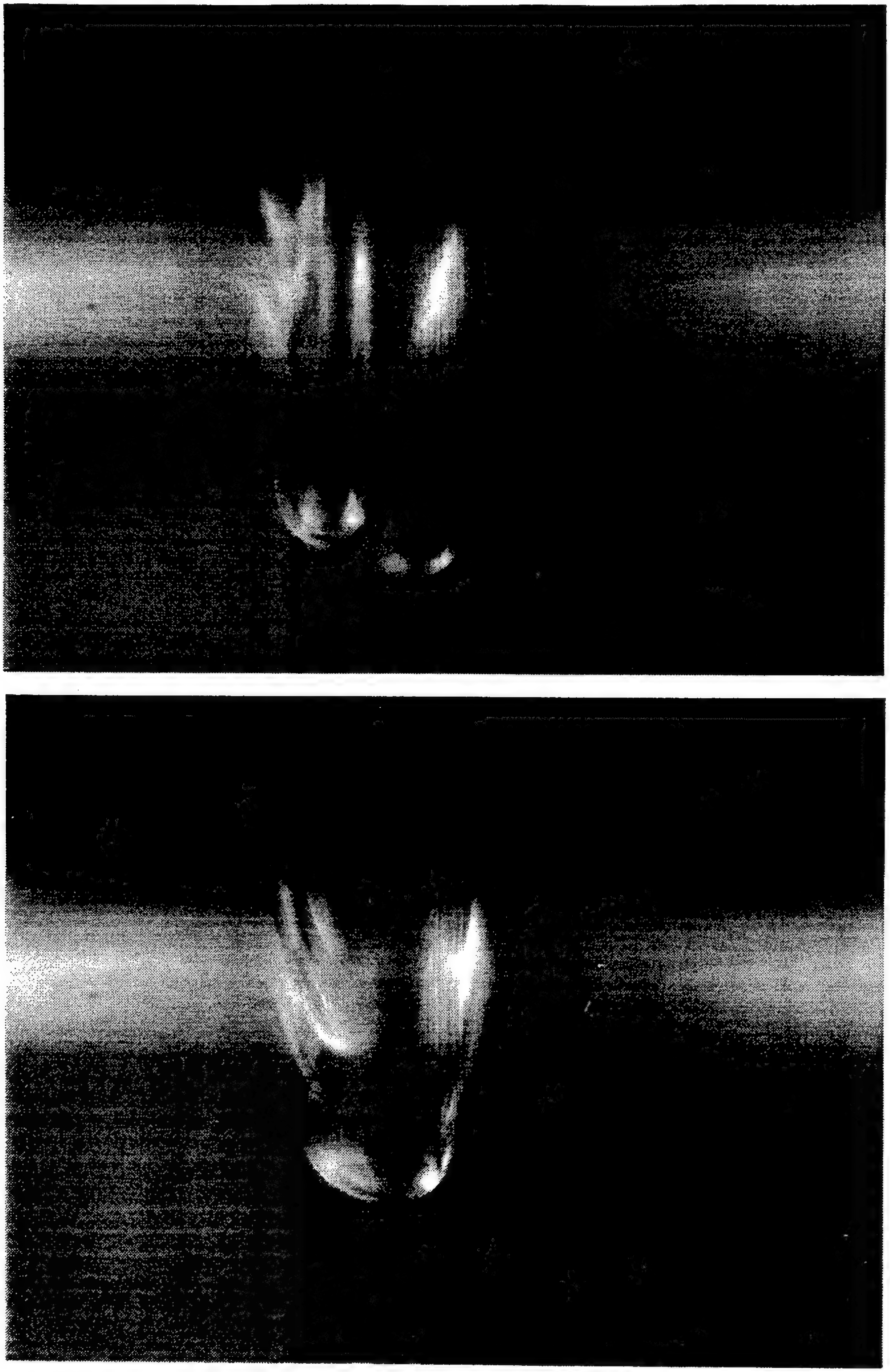
(c): Raman imaging shows non-uniform distribution of vaporized oxygen



Conceptual Representation of Experimental Results

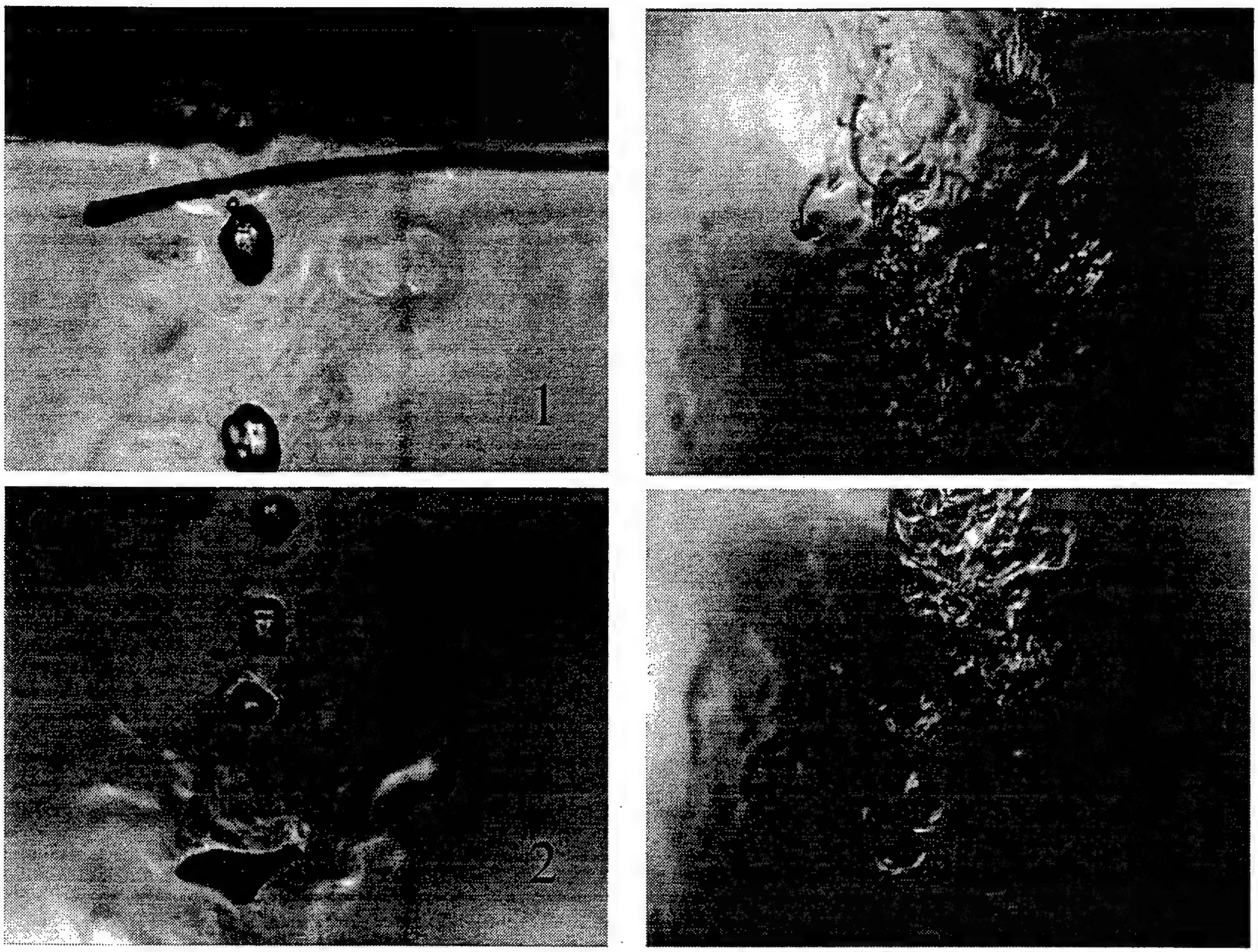


EXPERIMENTAL RESULTS:
Droplets



Supercritical Oxygen Drip through He into N2 @ 1100 psig

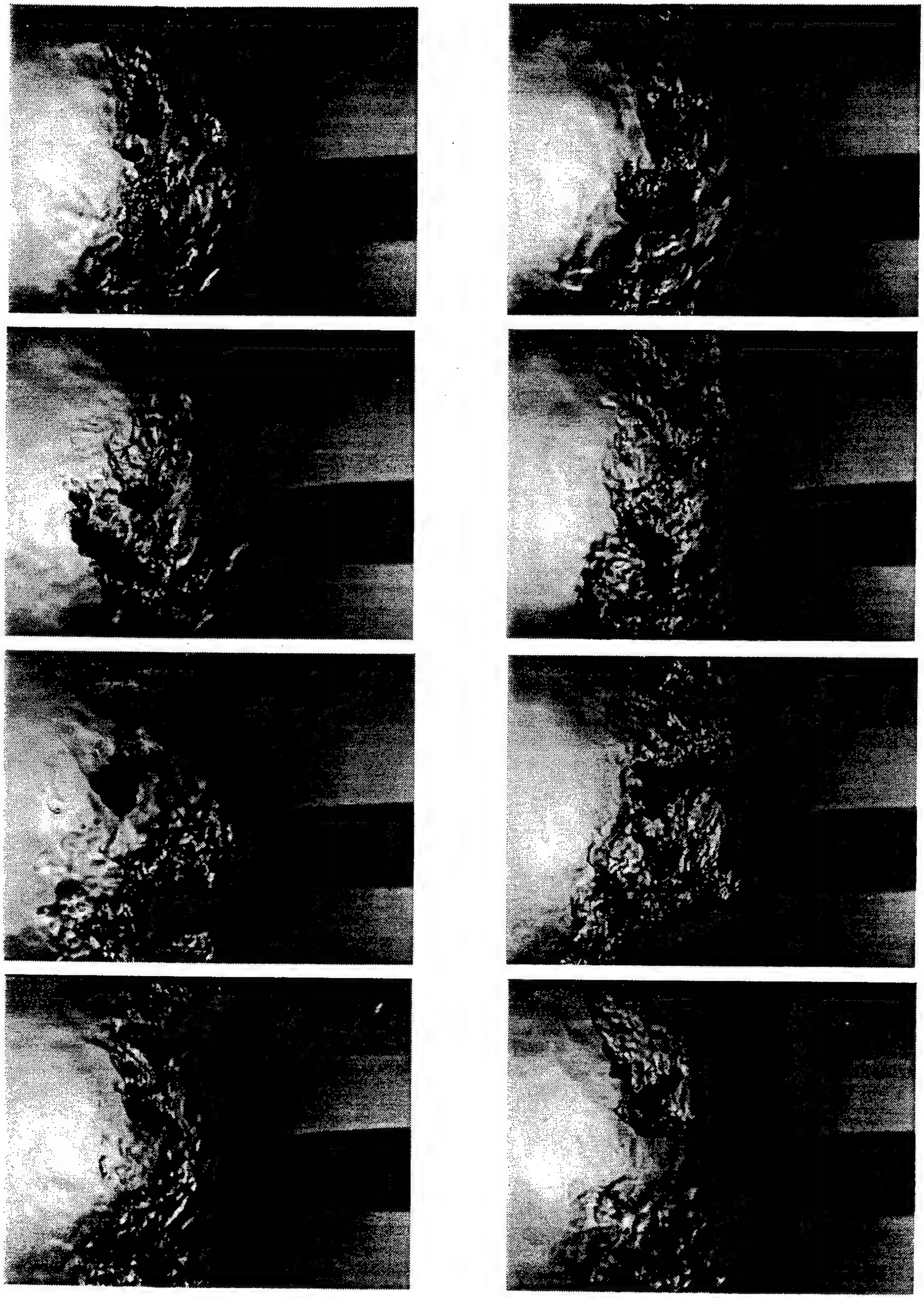
Transcritical Injection of Liquid Oxygen Droplets



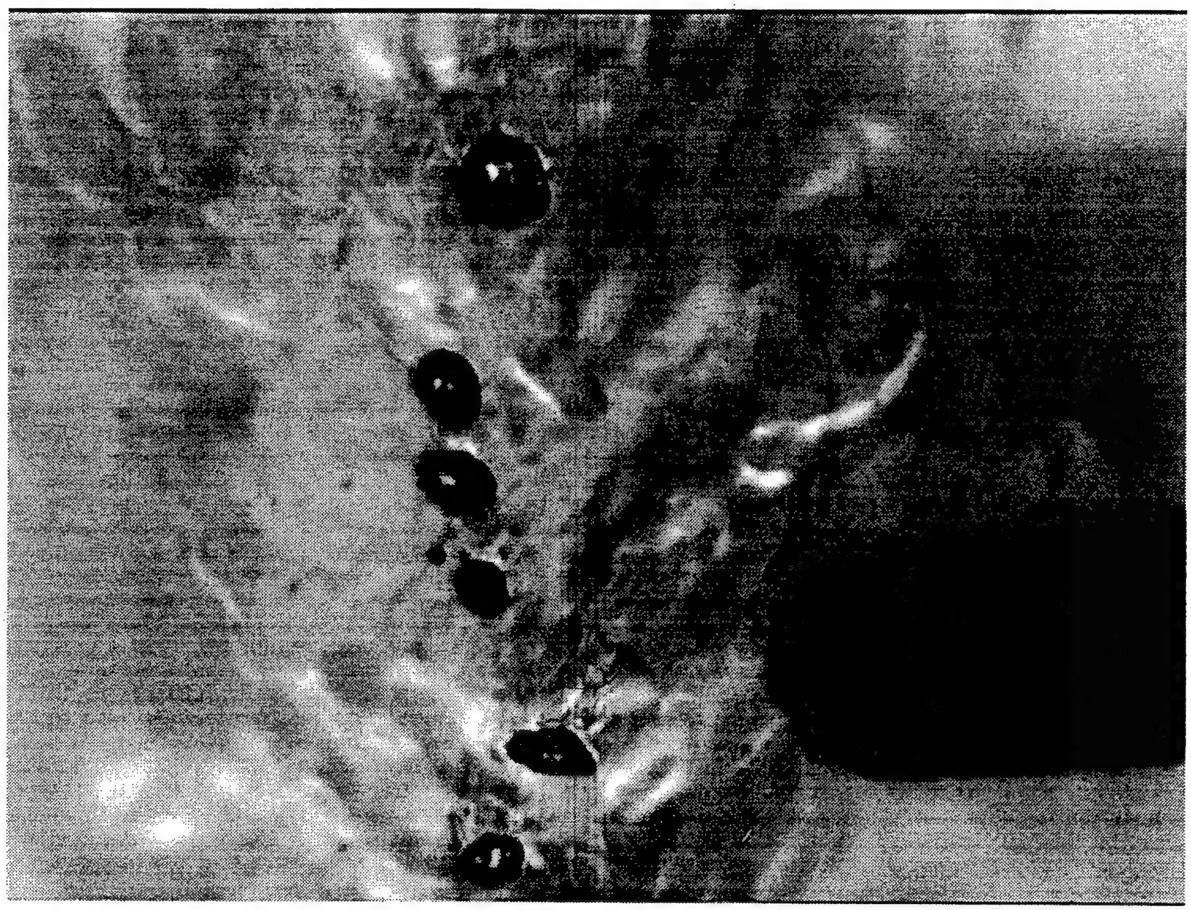
Ambient: N2 @ 970 psig,290K.
O2 Injector Temperature: 177K.

Drops formed in chilled helium, nitrogen flowing across stream.

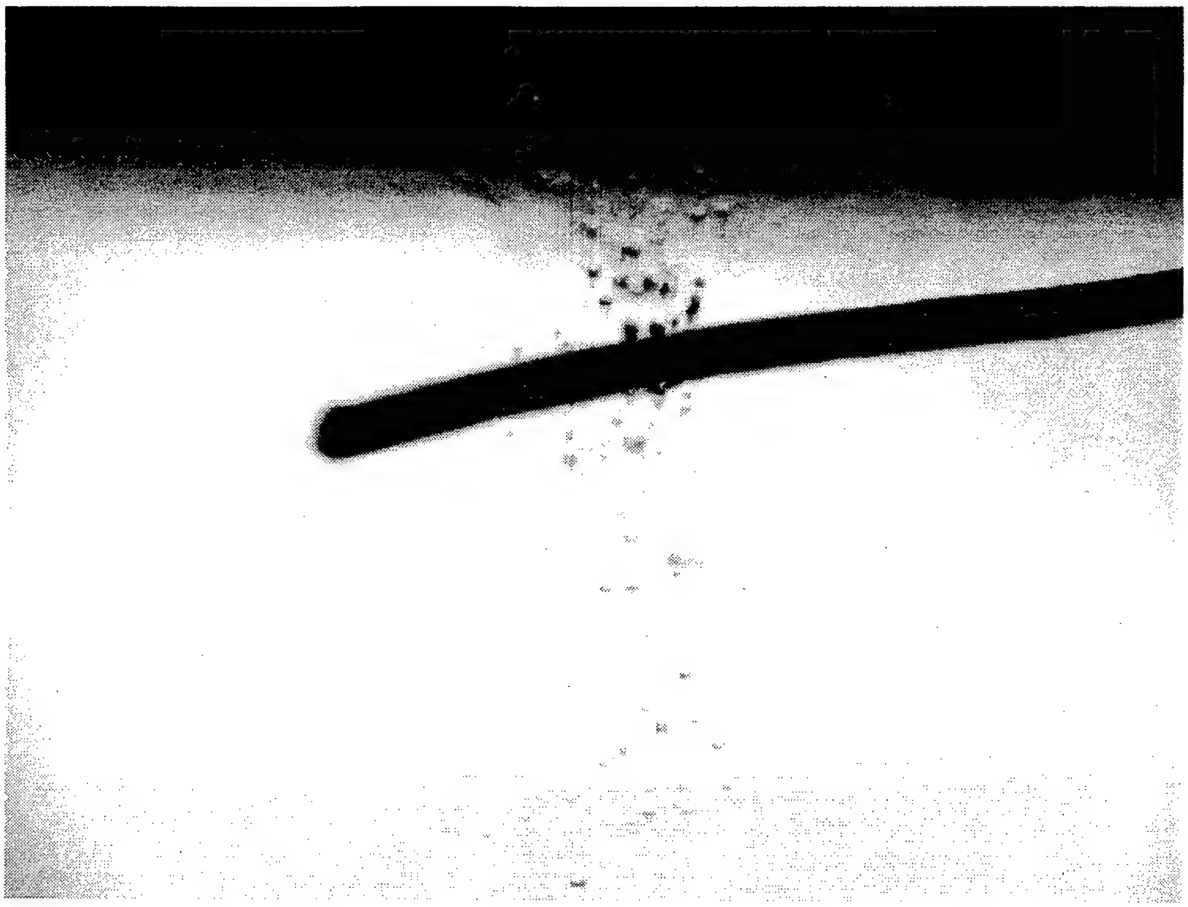
Sequence corresponds to increasing distance downstream.



Transcritical Oxygen Drops in N2 @ 970 psig



Oxygen Droplets Formed in He and Falling into N2
@ 1000 psig, 280K



Oxygen Droplet Condensation
in Helium @ 1000 psig, 280K

EXPERIMENTAL RESULTS:
Jets

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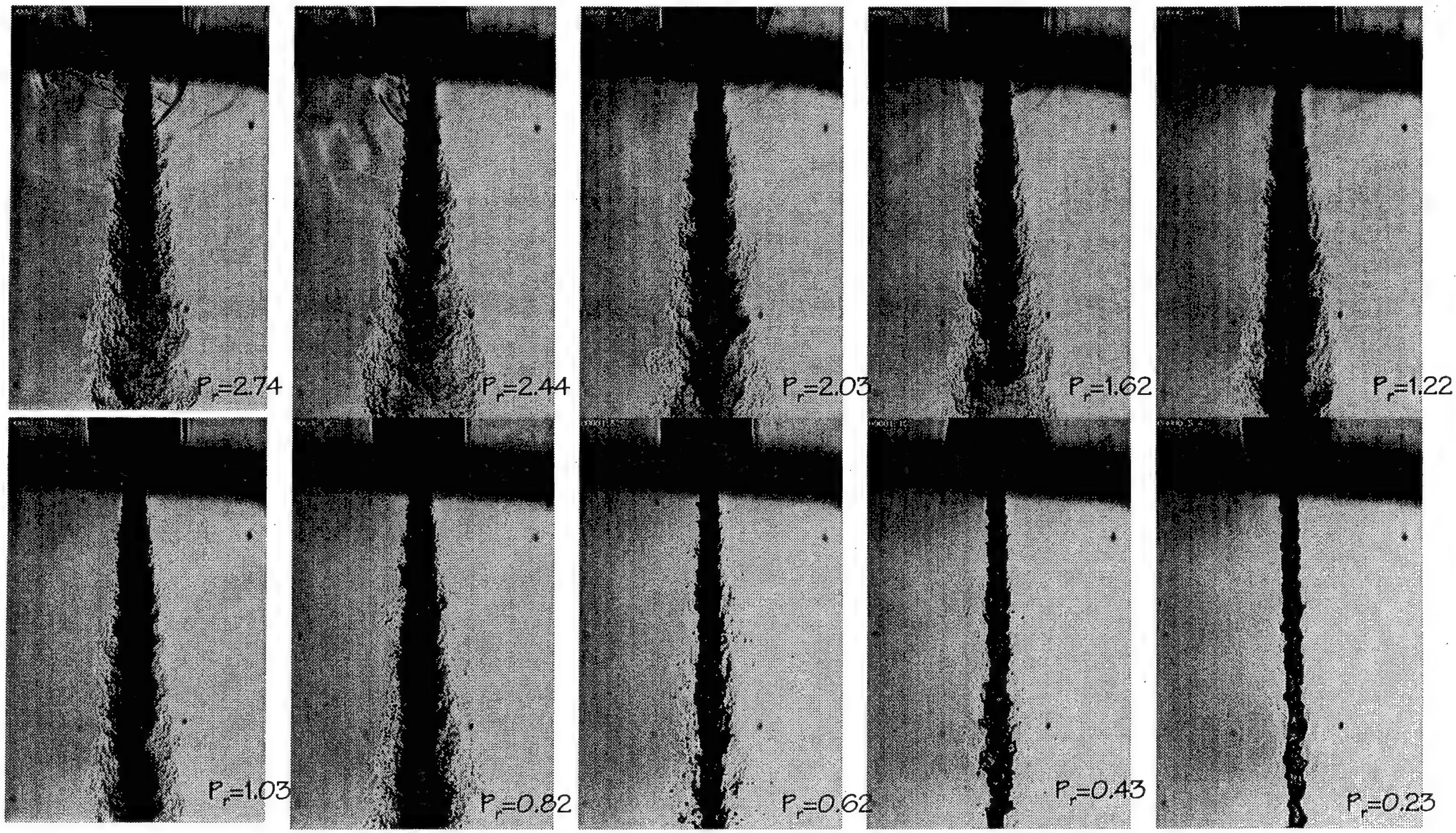
N_2 into N_2

Back-illuminated images of nitrogen injected into nitrogen at a fixed supercritical temperature of 300 K but varying sub- to supercritical pressures. $P_r = P_{ch}/P_{critical}$. Re= 25,000 to 75,000. Injection velocity: 10-15 m/s. Froud number: 40,000 to 110,000. Injectant temperature: 99 to 120 K.

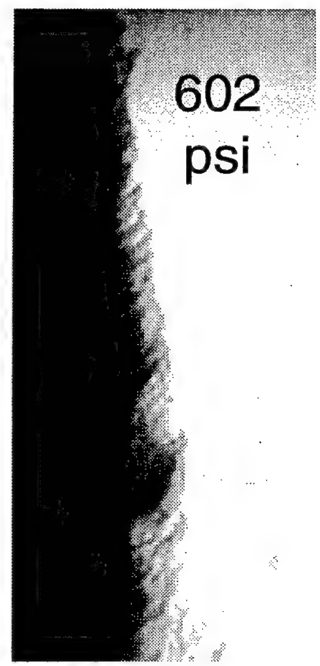


PRESSURE DEPENDENT MIXING LAYER STRUCTURE

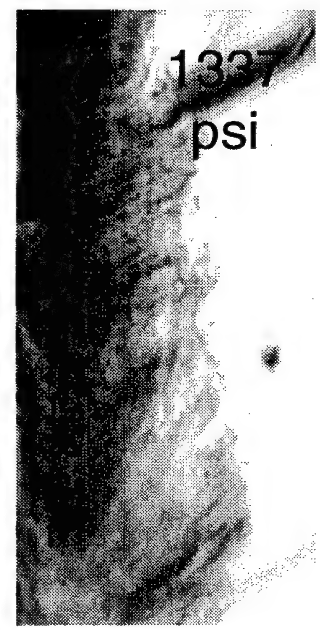
Nitrogen/nitrogen system (P_{cr} = 493 psi, T_{cr} = 126 K)

T_{inj} = 128 K, T_{amb} = 300 K, mass flow = 350 mg/s

Low Pres.
Subcritical
Droplets



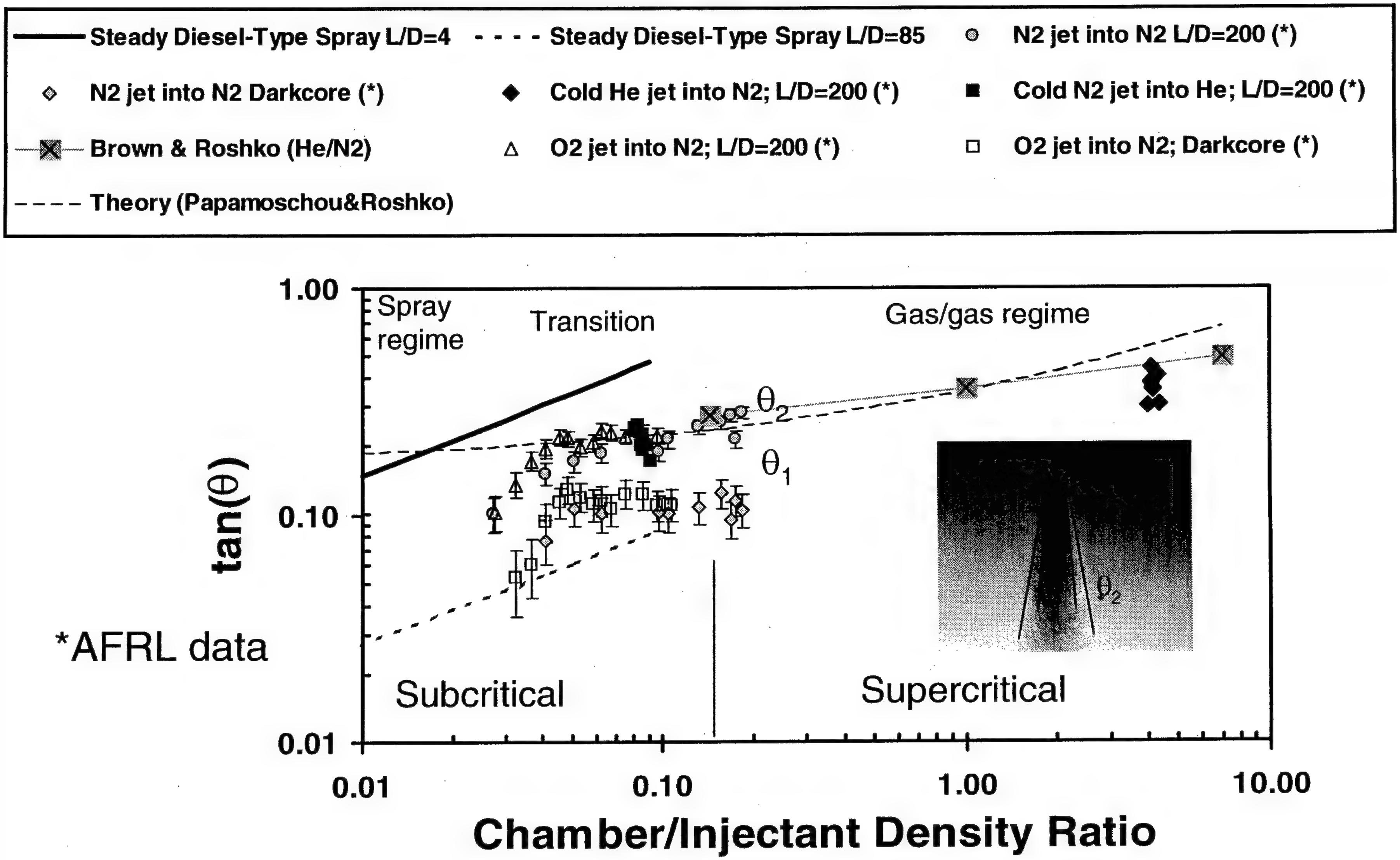
Mod. Pres.
Supercritical
Ligaments



High Pres.
Supercritical
Gas layers

Air Force Research Laboratory
(6.1)

Sub- and Super-critical Mixing Layer Physics



*AFRL data

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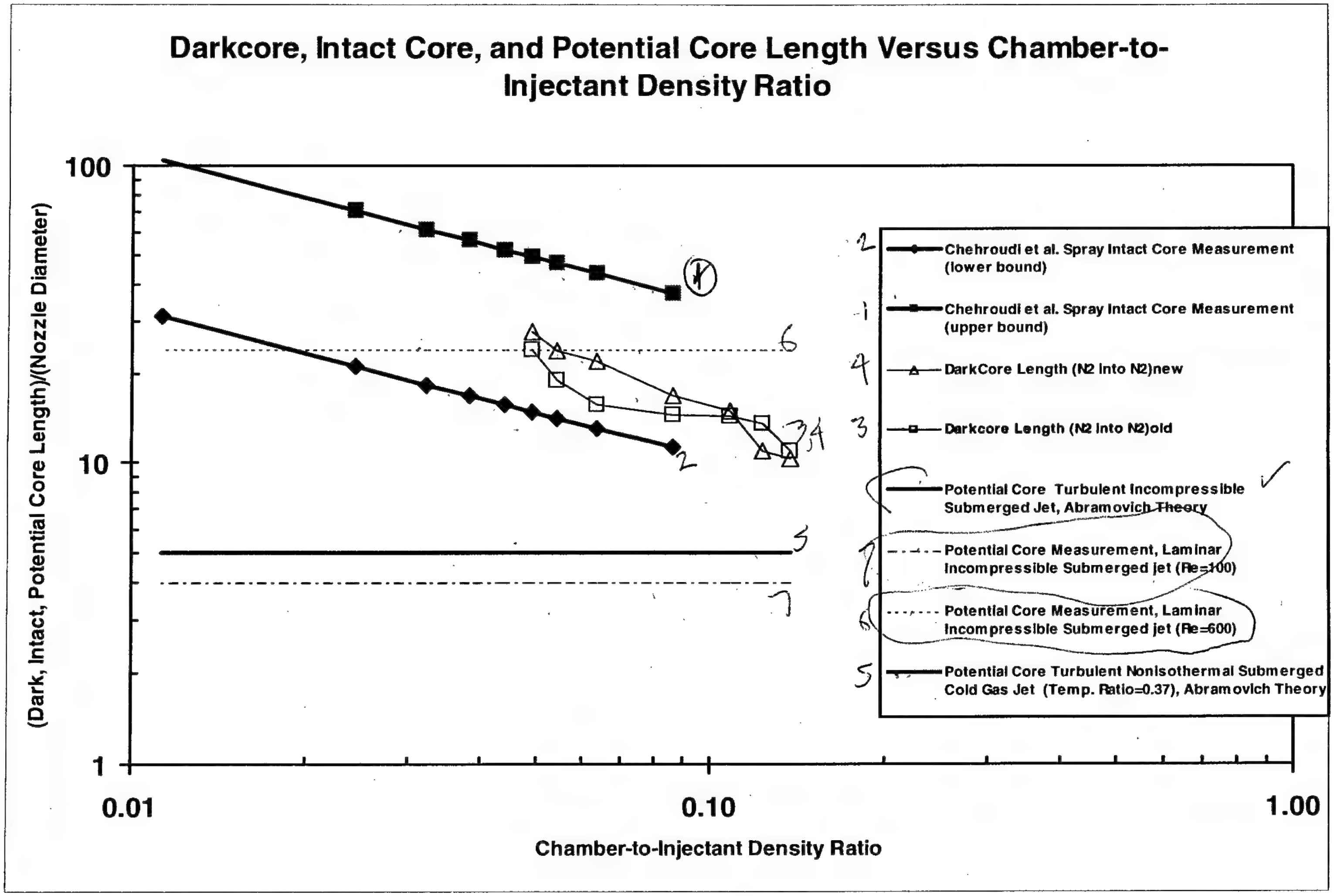
Jet Dark Core Length



Characteristic Times

- Characteristic bulge formation time (τ_b) at the jet interface (Tseng et al.): $(\rho_l L^3/\sigma)^{1/2}$; ρ_l, L, σ are liquid density, characteristic dimension of turbulent eddy, and surface tension, respectively.
- Characteristic time for gasification (τ_g) (D-square law): D^2/K; D and K are drop diameter and vaporization constant.
- A Hypothesis: If these two characteristic times (calculated for appropriate length scales) are comparable then an interface bulge may not be separated as an unattached entity (onset of the gas-jet behavior at supercritical condition)

Similar equation format for different cases

- Theoretical isothermal liquid spray growth rate (θ_s) based on Orr-Sommerfeld equation and stability analysis to find the wavelength of the most unstable interface wave:

  $$\theta_s \cong 0.27\,[0 + (\rho_g/\rho_l)^{0.5}]$$

- Papamoschou/Rashko theory for incompressible variable-density gaseous mixing layer/jet:

  $$\theta_{P/R} \cong 0.17\,[1 + (\rho_g/\rho_l)^{0.5}]$$

- Dimotakis theory for incompressible variable-density gaseous mixing layer/jet:

  $$\theta_D \cong 0.212\,[0.59 + (\rho_g/\rho_l)^{0.5}]$$

- ALL HAVE THE SQUARE ROOT OF DENSITY RATIO AND THE SAME EQUATION FORMAT

Proposed "intuitive/smart" equation

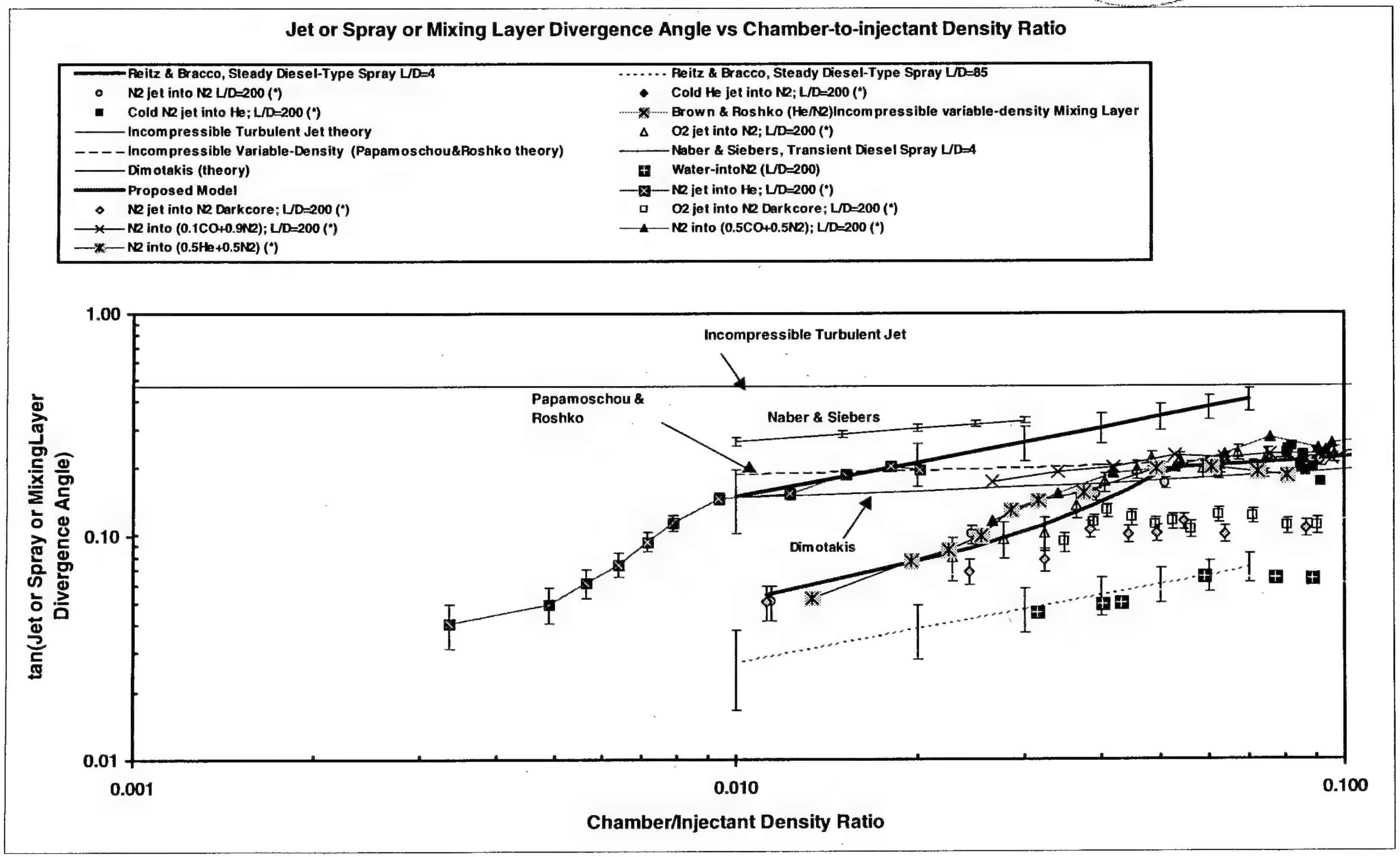
- Based of the information of the previous slide the following "intuitive/smart" equation is proposed for both sub- and supercritical measured growth rates:

$$\theta_{Ch} \cong 0.27\,[\,(\tau_b/(\tau_b+\tau_g)) + (\rho_g/\rho_l)^{0.5}]$$

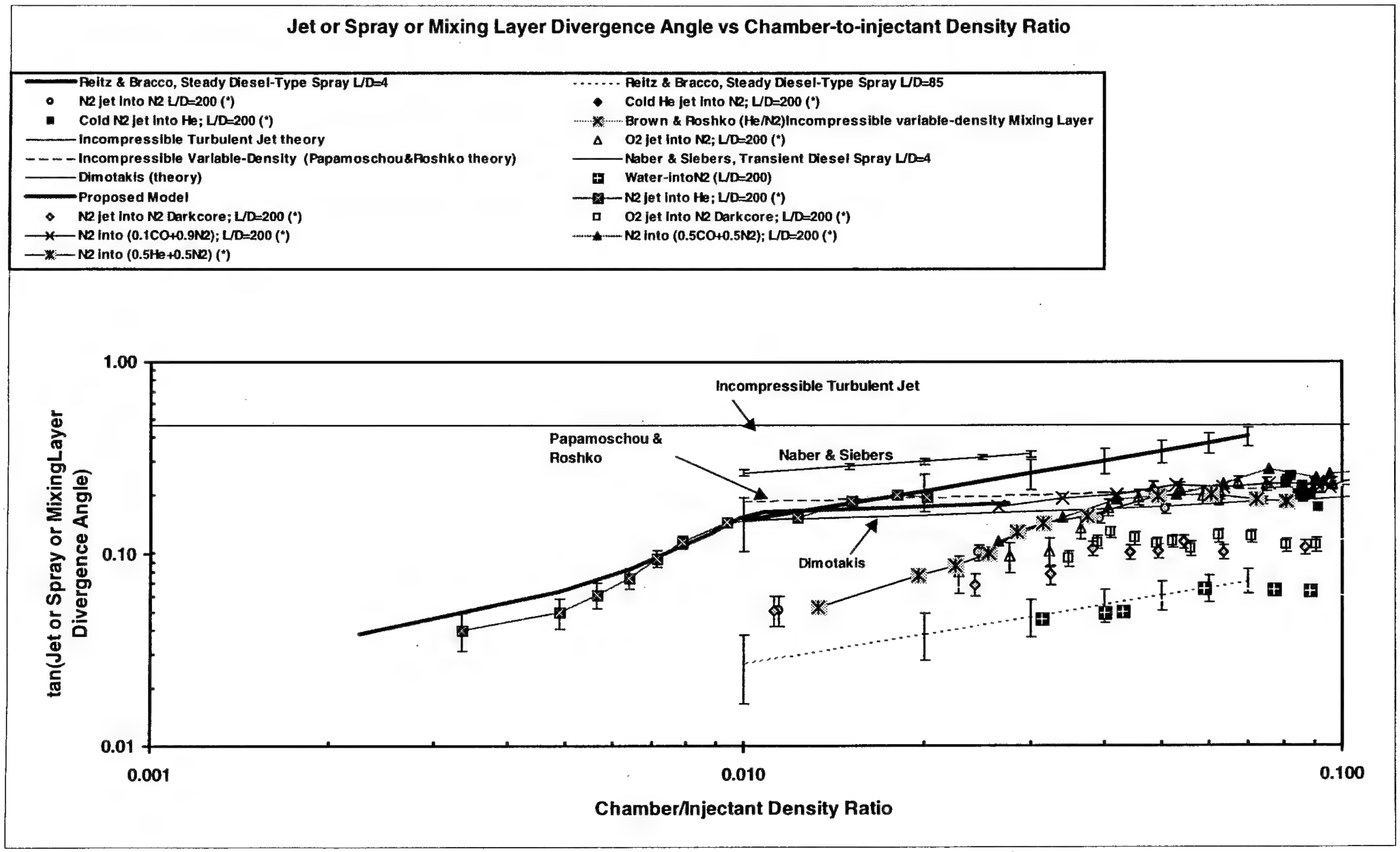
Note:

- For isothermal liquid case: $\tau_g >> \tau_b$ and $\tau_g \rightarrow \infty$. It then collapses to the isothermal spray case.
- For subcritical the $(\tau_b/(\tau_b+\tau_g))$ is calculated until it reaches 0.5. After that it is maintained constant at 0.5n for supercritical jet. The transition point is found to be approximately when $(\tau_b/(\tau_b+\tau_g)) \cong 0.5$.
- Only a horizontal axis variable transformation is needed to fit with the N_2-into-He experimental growth rate data.

Comparison of the proposed equation (solid red line) with experimental data



Comparison of the proposed "intuitive/smart" equation with the N_2-into-He experimental data



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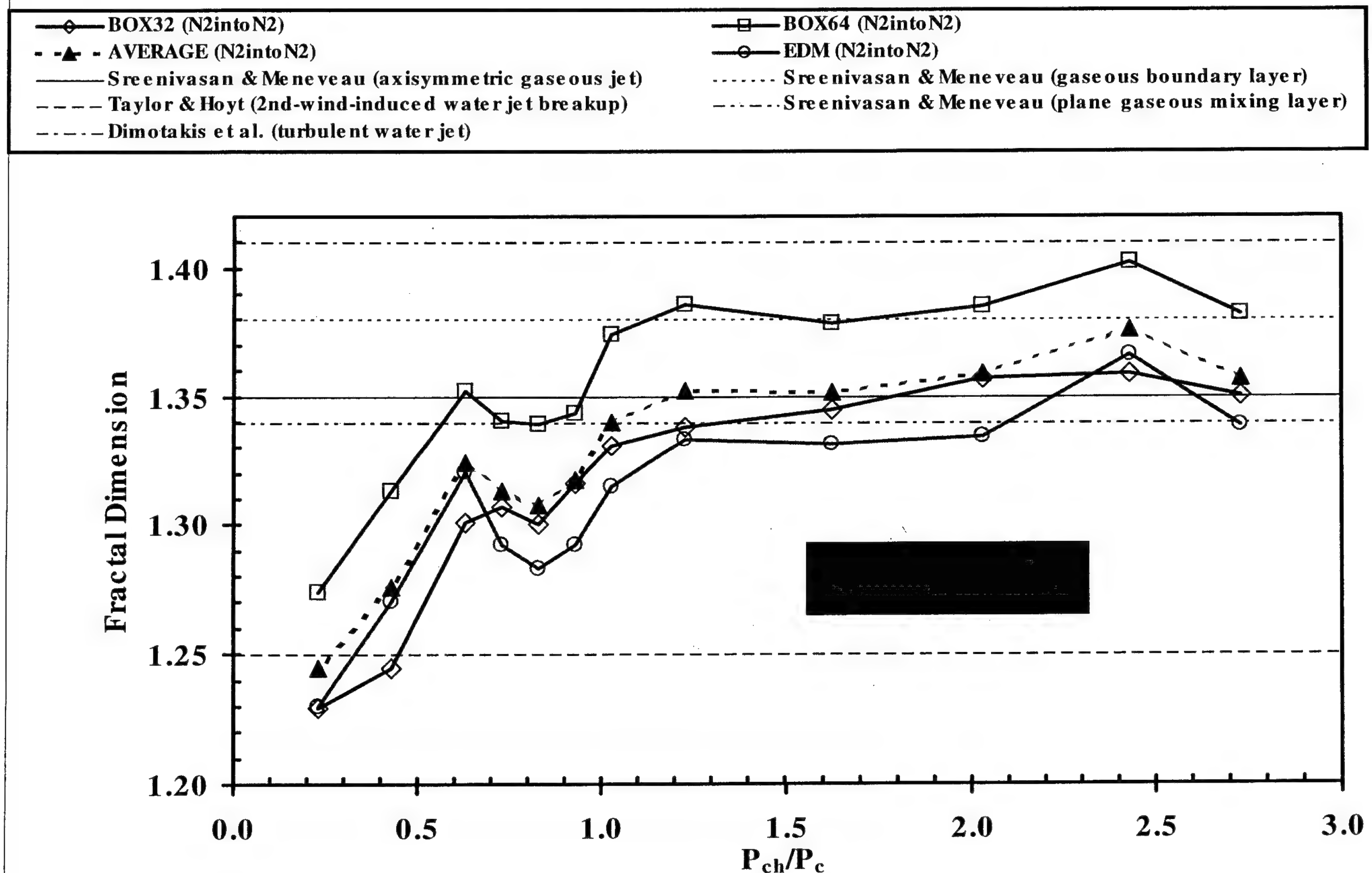
FRACTAL DIMENSION vs. RELATIVE PRESSURE



Instant images of Sub- and Supercritical impinging jets

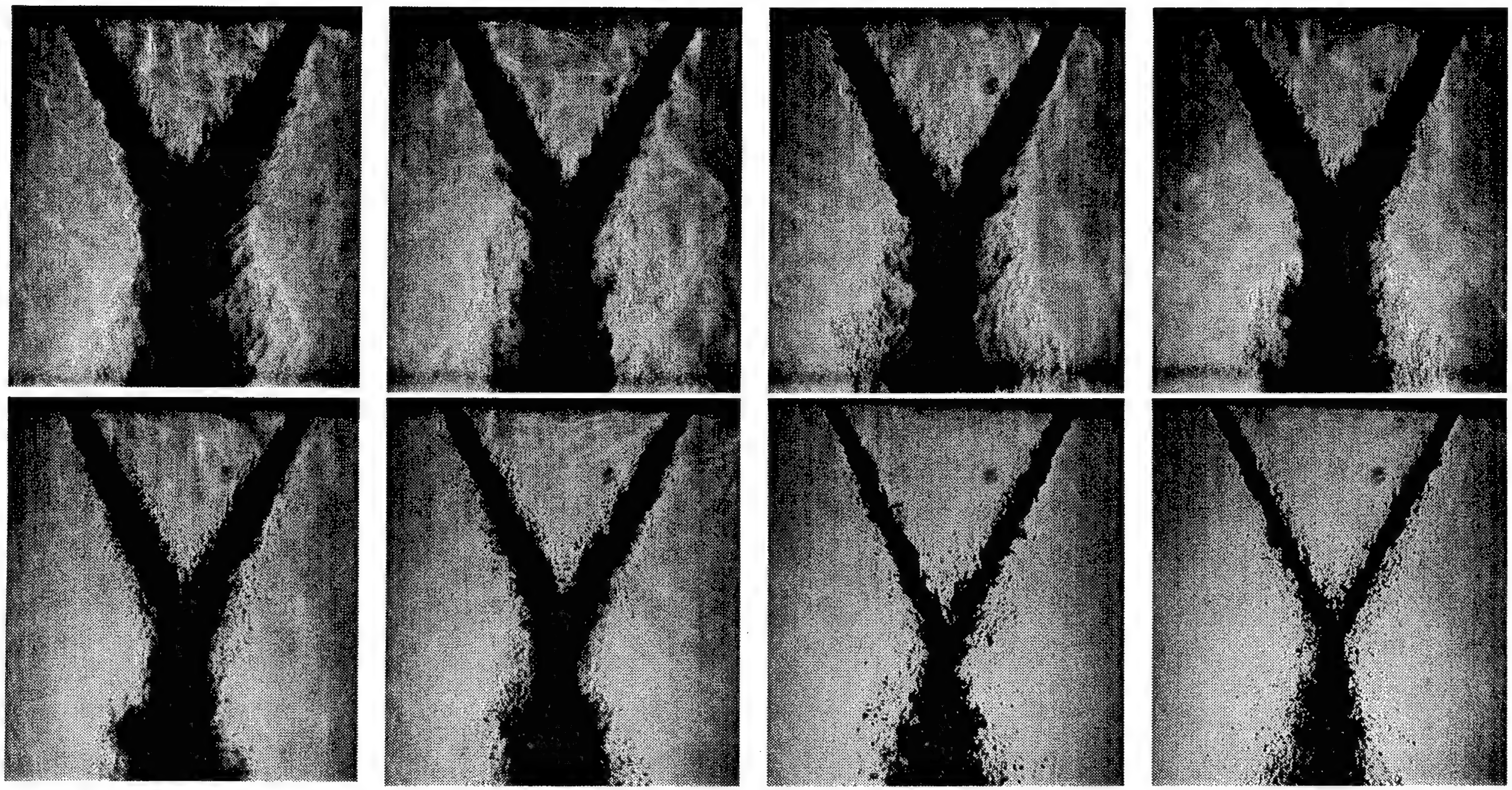
N_2 into N_2

($P_{critical}$ = 3.39 MPa; T_c = 126.2 K)

(P_{ch} = 800, 600, 500, 400, 350, 300,200,100 psig; from upper left to lower right)

(P_{ch} = 5.5, 4.2, 3.5, 2.8, 2.5, 2.1, 1.5,0.8 MPa; from upper left to lower right)

(Re=25,000 to 70,000; injection velocity:10-15 m/s)



Smooth/sharp/sharp--NIH; 60, 63, 1 - Image-Pro

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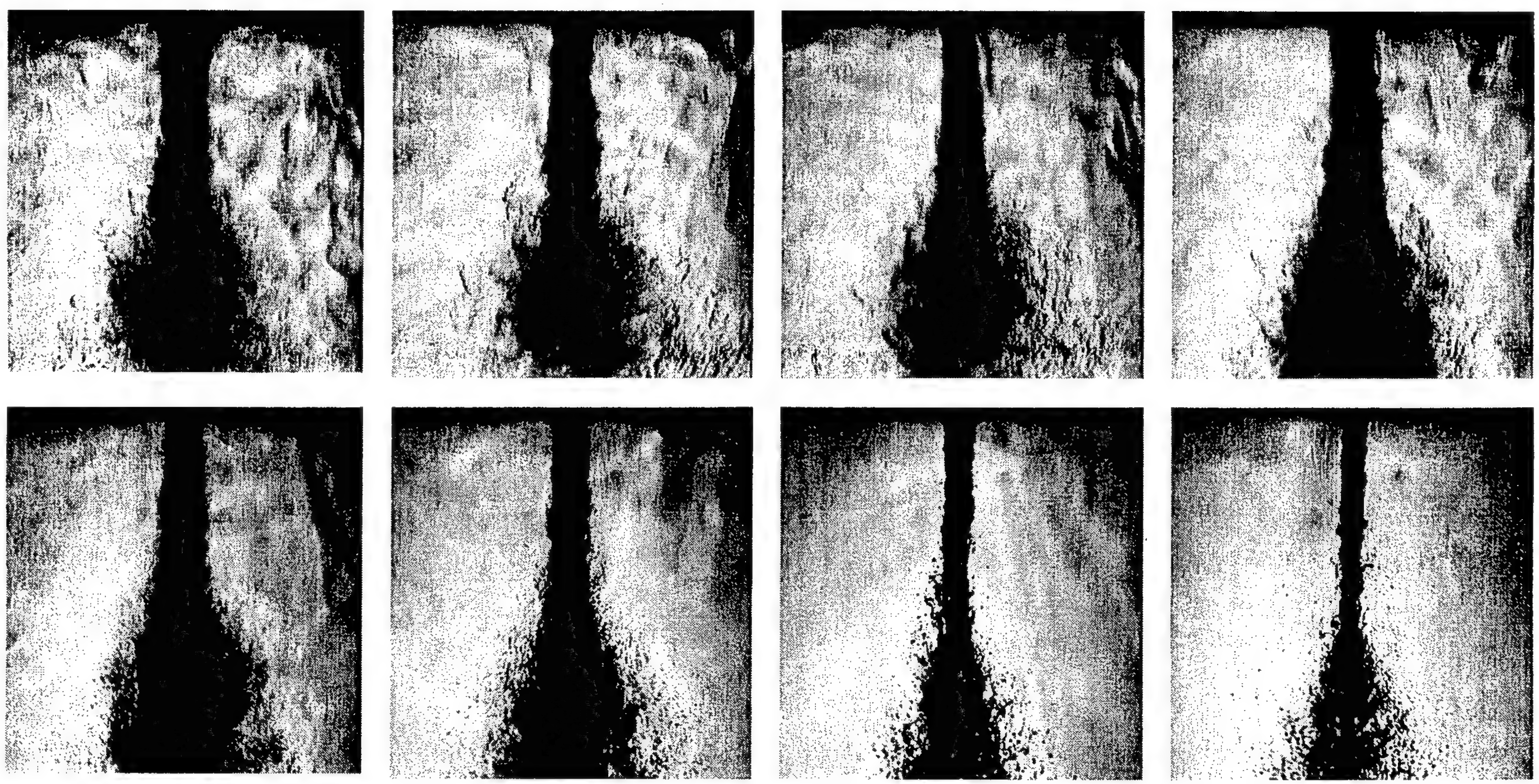

Smooth/sharp/sharp--NIH; 60, 63, 1 - Image-Pro

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Summary and Conclusions

- Cryogenic Jet structural differences are observed below and above the thermodynamic critical point.
- Liquid-Jet like appearance up to near critical point similar to second wind-induced liquid jet breakup regime.
- Gas-jet like appearance above the critical point. No drops are observed.
- A unique plot has been constructed for the jet growth rate covering density ratio range of up to a 1000.
- Measured growth rate (divergence angle) of our cryogenic jet under supercritical condition agrees well with the theoretical equations by Papamoschou & Roshko and Dimotakis for incompressible variable-density turbulent gaseous mixing layer.

method

q

Summary and Conclusions (cont.)

- Transition to full atomization regime is inhibited due to vanishingly small surface tension and heat of vaporization.
- All tend to strengthen the position that jets under the condition investigated here exhibit gas-jet like behavior at near and above the critical point of the injectant. Here the gas-jet like behavior is quantitatively demonstrated and verified for the first time.
- The relevancy of current injection models and some drop vaporization/combustion results under the conditions where gas-jet like behavior is detected should be reexamined.

Basic Research Opportunities

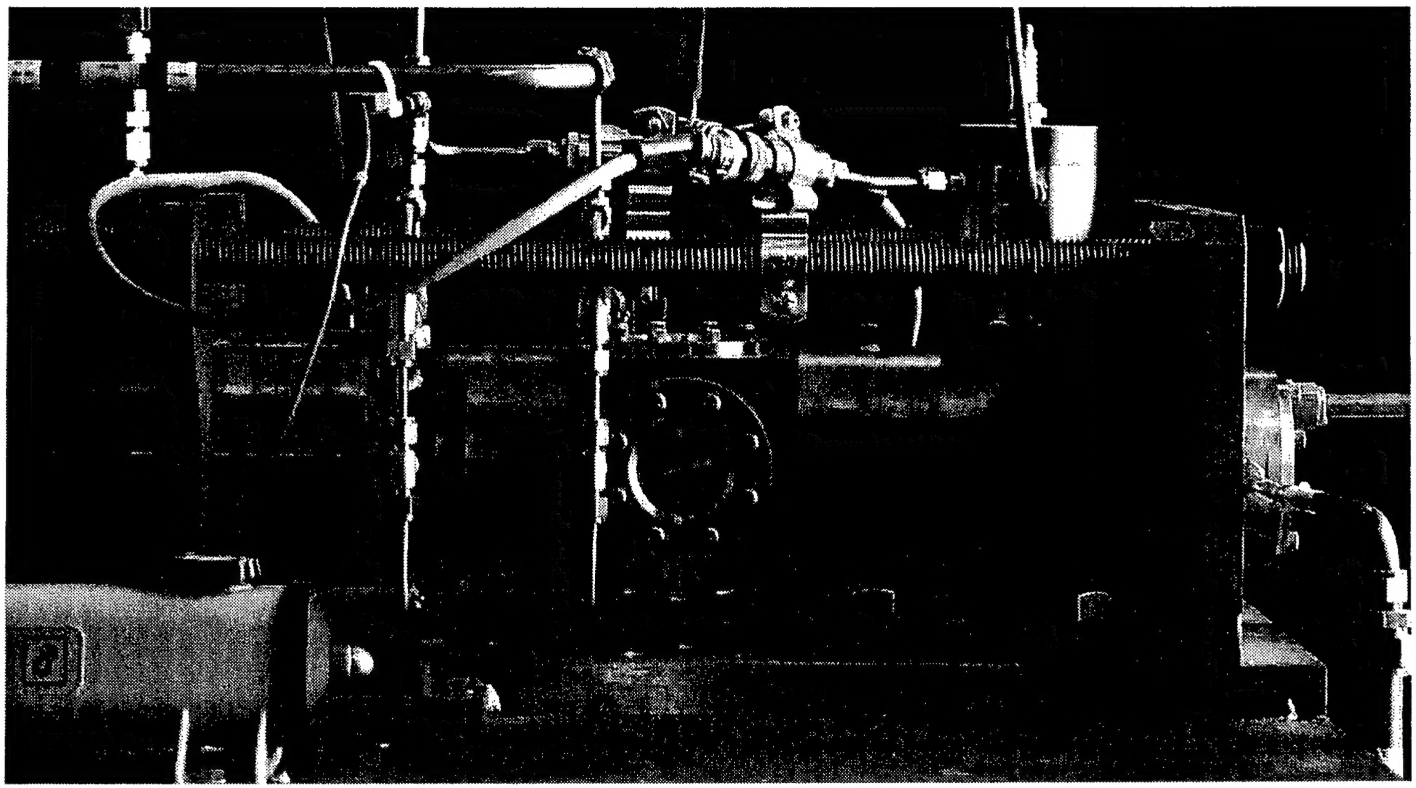
1990's at AFRL

- High pressures, "steady."
- Gas/gas injection

Future

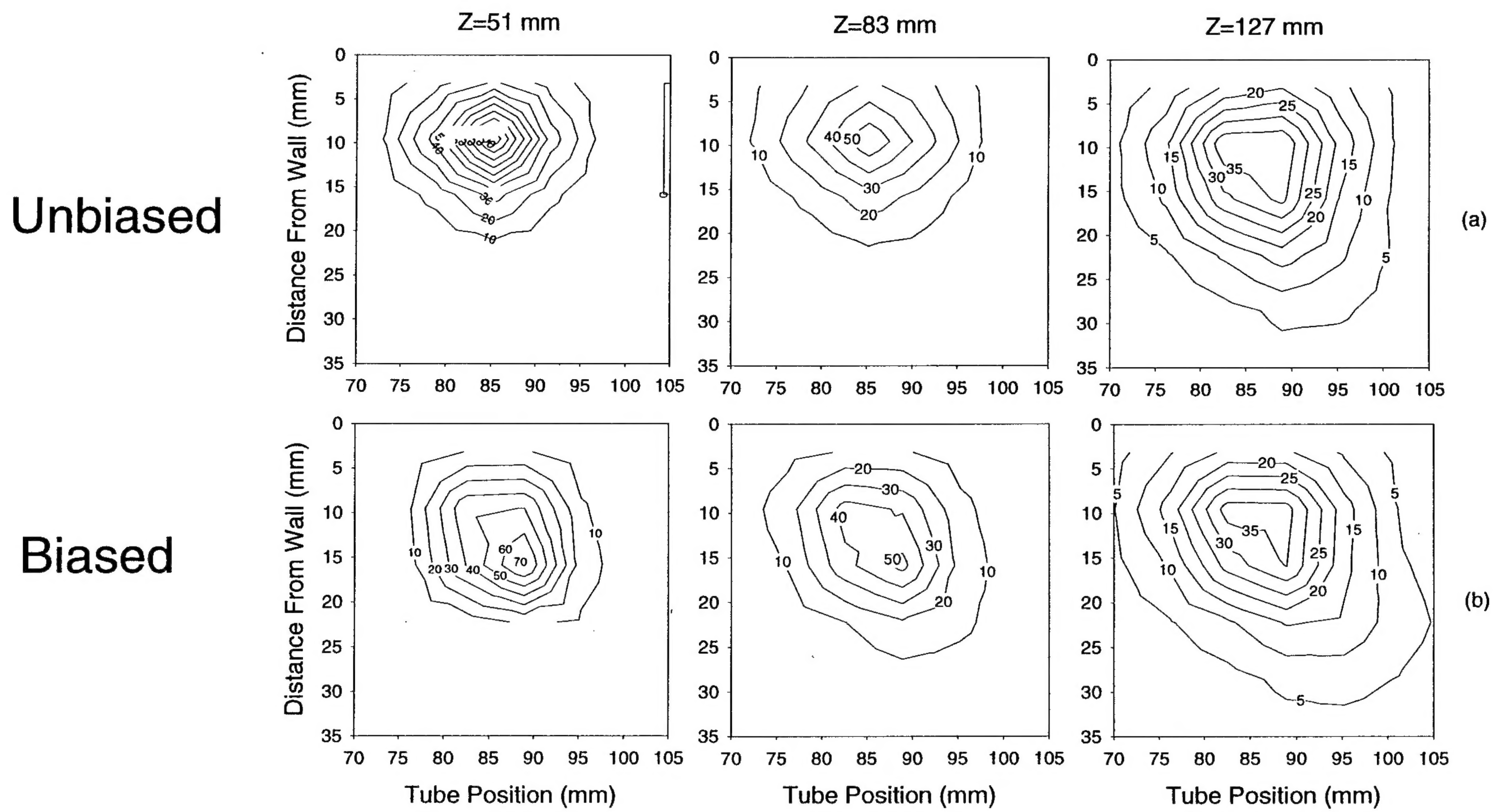
- Injector/chamber interactions
- Transients
  - Organized
  - Un-organized
- Revolutionary cycles
  - Pulsed detonation propulsion
  - Combined cycle

Gas/gas hardware



LOX post biasing study

Water/N2, density ratio = 117



Pulsed Detonation Rocket Engine Test at AFRL